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THESIS

INFRARED DETECTION OF SURFACE VEHICLE,
CALCULATION USING ATMOSPHERIC MODEL LOWTRAN 6

by

Ioannis Egolfopoulos

December 1984

Thesis Advisor:

A. W. Cooper

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Infrared Detection of Surface Vehicle, Calculation Using Atmospheric Model LOWTRAN 6

bу

Ioannis Egolfopoulos Lieutenant, Hellenic Navy B.S., Hellenic Naval Academy, 1975

Submitted in partial fulfillment of the requirements for the degree of

MASTER OF SCIENCE IN ENGINEERING SCIENCE

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ABSTRACT

The thermal signature of a surface vehicle, "SPRUANCE" type destroyer, quantified by the signal to noise ratio of a thermal seeking missile head is parametrically investigated. The effects of the ship internal temperature, ship body paint emissivity, sky condition, sun elevation angle above the horizon, atmospheric profile, missile optics and flight altitude are examined in detail. Results show that both the ship body temperature and signal to noise ratio increase as the incident solar energy and the ship body paint emissivity increase and that the signal to noise ratio appears a peak for sun elevation angle in the range of 40° to 60°. Moreover the signal to noise ratio increases as the missile flight altitude decreases and keeping the other parameters constant higher values are found for the Midlatitude Summer atmospheric profile than for the Tropical atmosphere.

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I. INTRODUCTION

The basic function of a thermal-seeking-head missile is to track on a surface vehicle using the vehicle's radiant intensity. This operation is based on the fact that a moving vehicle shows a generally higher surface temperature that the environment. Reasons for this are the existence of operating engines, resulting in both increased engine room temperature and "hot" combustion products exiting into the atmosphere from the stack, the operation of a number of devices which result in heat generation, the requirement for certain compartment temperatures comfortable to the crew, and the incident solar energy on the painted metallic vehicle surfaces.

Exact information about the heat seeking missile characteristics is not available because of their classification. Hence our study will be a parametric one in order to examine the relative influence of a number of parameters on the vehicle thermal signature which is quantified by the signal to noise ratio of the thermal detecor head. The range of the parameter values will be chosen using both experience and unclassified information.

Exceptional cases such as operation of the vehicle close to the coast or other vehicles, which could have similar or higher temperatures, will not be considered. At this point we must realize that the thermal interaction of the vehicle with the environment is a very complicated phenomenon, and its exact determination for the different parts of the ship body is outside of the scope of the present work. Instead we will consider a simplified case of uniform interaction for all the ship parts with the environment (convection, radiation, solar absorption energy) and will limit the source for

the thermal head detector environmental noise only to that of the sea water.

The main procedure steps of the present study will be :

To create a simplified model for the calculation of the vehicle body temperature and to show what parameters affect it and in what direction.

To determine the "hottest" vehicle area and combining with the operating characteristics of the missile to calculate the dimensions of the tracked area of the vehicle.

To calculate the total radiant intensity of the tracked area of the vehicle in a certain wavelength range, where the thermal detector head is operating.

To calculate the atmospheric transmittance, using LOWTRAN 6, for each given combination of the parameters of our interest in the given detector wavelength range and consequently the radiant heat flux detected by the thermal head.

To create an appropriate detector model that could show optimum operation for the conditions of our problem, and finally to calculate the creating signal-to-noise ratio which quantifies the vehicle thermal signature and describes the detector operational quality.

To discuss the relative influence of all the pertinent parameters and to propose certain considerations about both the vehicle and the missile.

Without expecting significant quantitative importance for our results, due to the simplifications that have been made, their qualitative importance must be more remarkable.

Parameters that will play significant roles in our study will be:

vehicle internal temperature, vehicle body paint emissivity, sky condition, sun elevation angle above the horizon, atmospheric profile,
missile flight altitude,

instantaneous Field of View (I.F.V.) created by the detector's optic system,

detector material,

The present work consists of five Chapters. In the first Chapter we state the assumptions of the problem, we calculate the vehicle body temperature, and we determine its "hottest" area which is tracked by the detector. In the second Chapter we use the LOWTRAN 6 program in order to determine the atmospheric transmittance and in the third Chapter we calculate the total vehicle and background radiant intensity incident on the tracker aperture. In the fourth Chapter we calculate the detector signal-to-noise ratio for all the possible combinations of the parameters, and finally in the fifth Chapter we discuss the relative importance of each parameter and we suggest ways that could improve both the vehicle thermal signature and the detector operation.

II. SHIP MODELING

In order to create an exact ship model, it is necessary to acquire a significant amount of empirical data. For the present study, and for several reasons, the available data are limited, and are based only on the unclassified literature given in the first six references, with relatively reasonable assumptions.

A. ASSUMPTIONS

All compartments of the ship have a uniform temperature of 20°C=293.15K except the main engine rooms 1 and 2, which have a temperature of 30°C=303.15K.

The stack exit temperature is uniform and we study the heat radiation from the exit plane neglecting the plume radiation. Reasons for doing this are the non-uniformity of the plume geometry and temperature distribution, and the fact that it is cooled significantly in a relatively short distance from the exit plane [Ref. 3]. We choose as stack exit temperature $306\,^{\circ}F=152.2\,^{\circ}C=425.4K$

In order to account for the heat convection from the ship to the air we will use the relative wind-speed of 30 Knots. For this case we will have forced convection with a heat transfer coefficient in the range of 5 to 10 (BTU/H FT^2R) or equivalently 28.4 to 56.8 (W/M 2K). We will not account for the heat leakage from the ship body to the sea water or for the heat loss due to the vaporization effect on the wet surfaces above sea level. For this reason and in order to be closer to reality we choose for our heat convection losses calculation the value of 10 (BTU/H FT^2R) or 56.8 (W/M 2K) as representative for the convection heat transfer coefficient.

The present study will be in Mediterranean atmospheric conditions and more specifically for the area of the Aegean Sea. We will use two atmospheric profiles i.e. Midlatitude Summer for months October through April and Tropical for months May through September. The corresponding ambient temperatures are 21°C=294.2K and 26.6°C=299.7K respectively [Ref. 6].

B. GEOMETRY AND MATERIAL OF THE SHIP

In Figures 2.1 and 2.2 you will find the ship side and top view respectively with the appropriate dimensions.

The material of the ship lower body construction is constructional carbon steel with thickness 13.5mm and thermal conductivity 54 (W/M K) [Ref. 9].

The material of ship body superstructure is duralaminum with thickness 3mm and thermal conductivity 164 (W/M K).

These small values of the thickness and the high-values of the thermal conductivities for both materials result in negligible temperature gradients inside the materials. Hence it is reasonable to assume a uniform temperature throughout the thickness of both materials.

Both structures are painted with gray paint i.e. the emissivity and absorptivity will be spectrally independent and equal between them. A typical value for oil paint emissivity - absorptivity is 0.94 [Ref. 10:p.07] and for TiQ gray paint is 0.87 [Ref. 4:p.278] both for temperatures 300K. From these values we observe that the available data give us paint emissivity - absorptivity values in the range of 0.90. Based on the facts that these values play a significant role in the determination of the ship body temperature and that intensive research resulted in the production of new paints with half the values of the already existing ones [Ref. 7] we will use two (2) different values i.e. 0.80,

0.55. Therefore we cover the useful range of the existing and the newly developed paints. At this point we must note that we don't have any available information about selectively absorbing paints that could exist and give the minimum possible thermal signature and our assumption about the equality between emissivity and absorptivity is inevitable as a first approximation. Moreover the solution of the problem for two (2) different emissivities constitutes a good research point for the relative influence of this factor.

The emissivity of the interior surface of the ship body will be approximated as that of stainless steel which is 0.44 [Ref. 4:p.278].

C. SOLAR HEAT FLUX IN THE SEA LEVEL SURFACE

In Figure 2.3 you will find the illuminance levels on the surface of the earth due to the sun for three different sky conditions i.e. Unobscured Sun, Sun with Light Clouds and Sun with Heavy Storm Clouds as a function of the elevation angle above the horizon. These three sky conditions and the elevation angle above the horizon will be two more parameters in the solution of our problem.

Since the given values in Figure 2.3 are in photometric units (Lux) and we want to convert into useful radiometric units (W/M^2) we have to determine a conversion factor.

We know that the maximum incident solar flux on the earth surface is approximately 900 (W/M^2) [Ref. 10:p.11]. This value coresponds to 1.15•10 5 (Lux) and therefore the conversion factor will be:

 $CF = 900 / 1.15 \cdot 10^5 = 5.826 \cdot 10^{-3}$ (W/M²Lux)

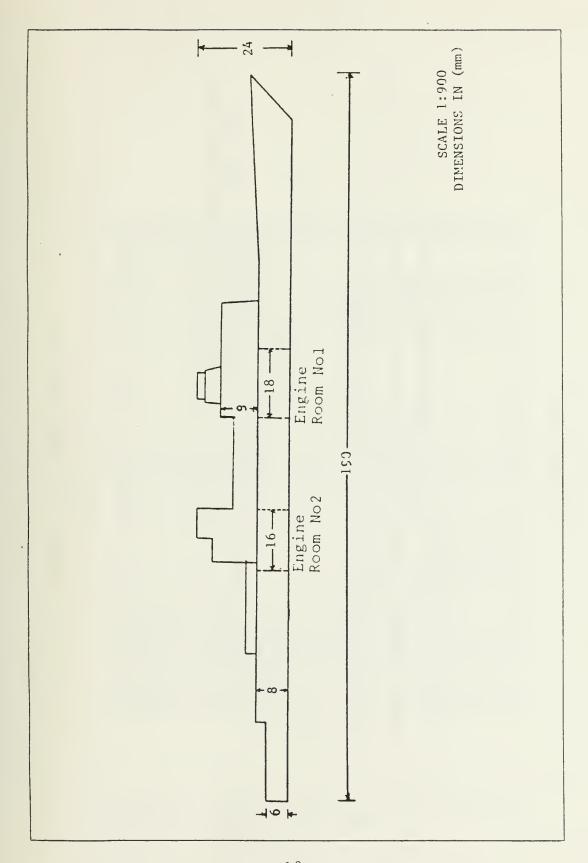


Figure 2.1 Ship Side View

Figure 2.2 Ship Top View

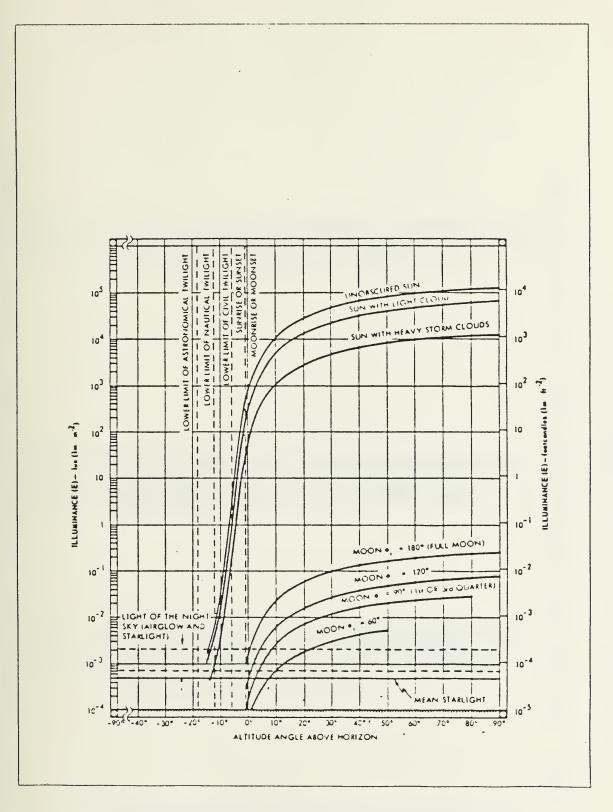


Figure 2.3 Illuminance Level on the Surface of the Earth [Ref. 5]

TABLE 1

Normal Component of Solar Heat Flux on a Surface at Sea Level

SUN WITH HEAVY STORM CLOUDS	33333333333333333333333333333333333333
SUN WITH LIGHT CLOUDS	11111111111111111111111111111111111111
UNOBSCURED SUN	46806284000002468048260000048260 2849109866777665577801356 284974196420010002468260000048260 28497419642001566677766557780 28497419642000246804826000004820 28497419711111111111111111111111111111111
ELEV. ANGLE	HHHHHHHHHOVOVOVOVOVOVOVOVOVOVOVOVOVOVOV

EAV	ののておかののであわりありたとらのととのするのもととて1000gでののごののじゅんとのあったこののようしょうしょうしょうしょ おおんしし しょうしょう しょうしゅう
SUN WITH LIGHT CLOUDS	088642066288408864206788402468806 188642066288408864200628840246915 18864220222222222222222222222222222222222
UNOBSCURED SUN	0246802468086420246806720864208642020 0987660482716187209766917 09878210976432697388470369246917 098786048271618720976432097642097 098786048271618720976430 09878777777777777777777777777777777777
EV. ANGLE	

Using this conversion factor and the values of Figure 2.3 we obtain the normal component of the solar heat flux in (W/M^2) on a surface at sea level. Table 1 demonstrates the above values.

In order to calculate the values of the solar heat flux falling obliquely on a surface we must introduce the incident angle Θ and multiply the values of Table 1 by $\cos\Theta$.

For a vertical face the angle Θ is identical with the elevation angle above the horizon. For a horizontal face i.e. the deck in our case, the angle Θ is the complement of elevation angle above the horizon.

D. CALCULATION OF SHIP BODY TEMPERATURE

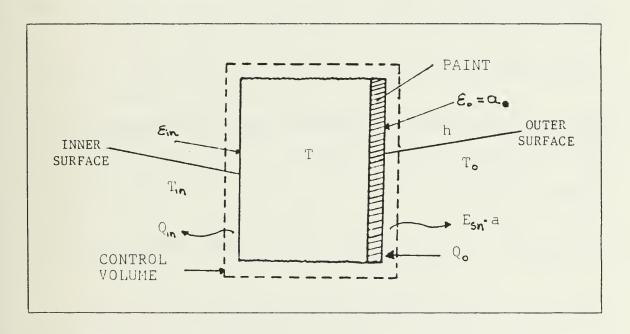


Figure 2.4 Control Volume for Determination of Ship Body Temperature

In Figure 2.4 you will find the appropriate control volume of the ship body and the pertinent energy flows that are related with it. For simplicity the control volume has

unit area exposed to both the environment and the ship interior. More specifically:

 ε_{\bullet} : paint emissivity

a: paint absorptivity (&= a)

 ε_{m} : emissivity of the interior surface. (ε_{m} = 0.44)

 T_{o} : ambient temperature

T_m: ship compartment temperature

Esn: normal component given by Table 2

Q_o: net heat exchange of control volume with the environment

 Q_{in} : net heat exchange of control volume with the interior of the ship

h: convection heat transfer coefficient with the air The absorbed solar heat flux is $E_{\mathsf{sn}} {}^{\bullet} a_{\mathsf{o}}$. The net heat exchange with the environment is due to heat radiation and heat convection i.e.

$$Q_{o} = \varepsilon_{o} \sigma (T^{4} - T_{o}^{4}) + h(T - T_{o})$$
(2.1)

The net heat exchange with the interior is due only to heat radiation i.e.

$$Q_{in} = \varepsilon_{in} \sigma \left(T^4 - T_{in}^4 \right) \tag{2.2}$$

Energy balance of the control volume under thermal equilibrium gives:

$$E_{sn} = \epsilon_{o} \sigma (T^{4} - T_{o}^{4}) + h(T - T_{o}) + \epsilon_{n} \sigma (T^{4} - T_{in}^{4})$$
(2.3)

or

$$F(T) = (\varepsilon_{\bullet} + \varepsilon_{in}) \sigma T^{4} + hT - \sigma(\varepsilon_{o} T_{\bullet}^{4} + \varepsilon_{in} T_{in}^{4}) - hT_{o} - E_{sn}^{\bullet} a_{o} = 0$$
 (2.4)

Equation 2.4 must be solved numerically. We will use the Steffensen Recursive Method [Ref. 11]. The idea of this method is that starting with an x close enough to the exact solution S of the equation F(x)=0 then the sequence:

$$x_{\mu\nu} = x_{\kappa} - [f^{2}(x_{\kappa}) / f(x_{\kappa} + f(x_{\kappa})) - f(x_{\kappa})]$$
 (2.5)

will converge very fast to the exact solution S.

In our case the equation is F(T)=0 and we are looking for the value of T that satisfies it.

In order to obtain an initial value so that our sequence will converge we use the value:

$$T_{in} = \left[\left(E_{sn} \circ a_o + \varepsilon_o \sigma T_o^4 + \varepsilon_n \sigma T_{in}^4 \right) / \left(\varepsilon_o + \varepsilon_{in} \right) \sigma \right]^{1/4}$$
(2.6)

The above initial value of T does not account for the heat convection and for our purpose gives acceptable initial values. Finally we stop the recursive formula when $|T_{k+1}-T_{k}|$ is less than or equal to 10^{-6} .

In Appendix A, Tables 5 through 18 demonstrates the ship body temperature for both horizontal and vertical faces, for all the combinations of two (2) atmospheric profiles, three (3) different sky conditions, two (2) internal (T_{in}) temperatures and two (2) paint emissivities.

E. DETERMINATION OF THE SHIP "HOTTEST" AREA

From the calculations in section D, we observe that the higher ship body temperatures happen to be in the region of the two engine rooms (higher internal temperature).

From the ship geometry, it is obvious that the dimensions of the engine room No 1 are greater than those of engine room No 2. Therefore the total radiant intensity (W/Sr) of engine room No 1 will be greater since for both regions we

have the same normal components of radiance (W/cm^2Sr) . In addition to that to each engine room there corresponds the same stack geometry also going a 'hot' region too.

Under this simplified assumption we conclude that for a heat seeking missile head the area of engine room No 1 from the stack exit all the way down to sea level is the most significant region.

At this point it is necessary to couple the ship with some characteristics of the missile detector operation.

Without getting involved in details of the detector operation, which will be studied in later sections, we will determine the detector Instantaneous Field of View (I.F.V.), so that we will know exactly the dimensions of the 'hottest' area which will be tracked by the detector and therefore we will be able to calculate the amount of radiant energy that will be transmitted to the missile.

F. DETERMINATION OF THE DETECTOR INSTANTANEOUS FIELD OF VIEW

The following range of data is based on our experience and research, aiming to calculate a reasonable and useful I.F.V. In Figure 2.5 you will find the corresponding geometry.

The range of the missile flight altitude will be from 50m to 100m above sea level. Since we want to reduce as much as possible the background noise, the detector must have an inclination with respect to the horizontal. We choose the range for this inclination (angle φ) from 2° degrees to 5° degrees.

From general cosideration for the detectors [Ref. 4] we know that I.F.V. is a function of the 'hottest' ship area that the detector has to track.

For the creation of I.F.V. it is necessary that the detector accepts radiation from inside an area which is related to an angle 2δ generated by the detector mirror system as in Figure 2.5. From the same figure we conclude the following:

 $\theta = 90^{\circ} - 6 - 8$

 $BC = AB + tan\Theta$

 $AC = (AB^2 + BC^2)^{1/2}$

CF = AC•sin6

 $CG = 2CF = 2AC \cdot \sin \delta = SIDE$

AF = CG/2tan6 = PATH

Where:

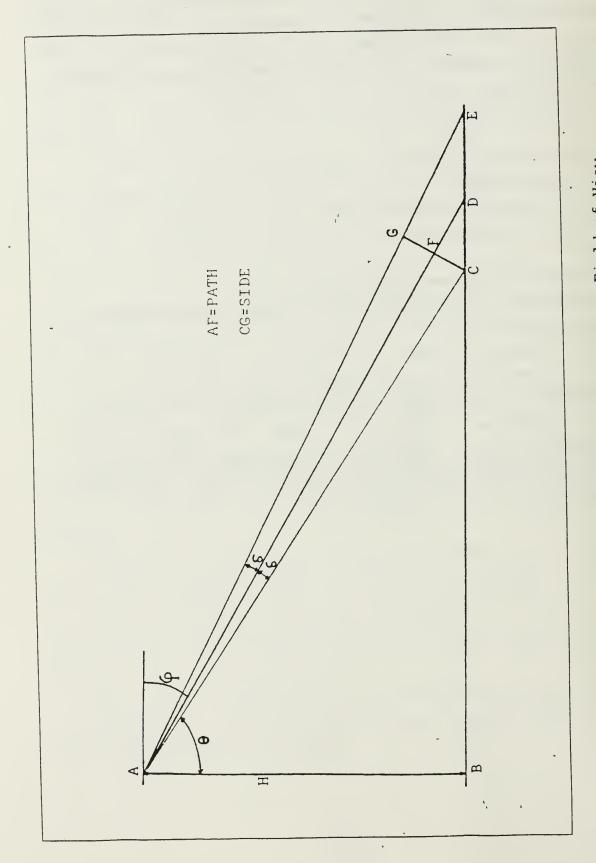
CG represents the side of the I.F.V.

AF represents the distance or path length between detector and the I.F.V.

AB = H : flight altitude

For simplicity we will consider that the shape of I.F.V. is square with side equal to CG of Figure 2.5

In Appendix B Tables 19 through 29 demonstrate the values of PATH and SIDE for all the possible combinations of φ , δ and H. Program SHIP 1, in Appendix E, has been used for the calculation of the above values.



Determination of Instantaneous Field of View Figure 2.5

G. DETERMINATION OF THE DIMENSIONS OF THE TRACKED SHIP AREA

In the previous section we have shown that the detector will track on the area of engine room No 1.

In Figure 2.6 you will find a three-dimensional sketch for the engine room No 1 area. Before we start the calculations of the different surfaces we will explain why the side of I.F.V. is greater than the width of the engine room No 1 which is $W_1=1,620$ cm. We consider as useful that the I.F.V. will include the whole area between sea level and the stack exit plane. The total height of this dimension is:

$$H_1 + H_3 + H_4 + H_5 = 720 + 900 + 360 + 180 = 2,160 \text{ cm}$$

From the Tables 13 to 24, we choose for each flight altitude and for constant $6=0.56^{\circ}$ the minimum SIDE that is greater than or equal to $2.160 \, \text{cm}$

Table 2 demonstrates the results of this choice

Based on Figure 2.6 we will now determine the characteristics of each individual surface.

 $A_{L} = W_{L} \cdot H_{L}$ Vertical face . $T_{lm} = 303.15K$

 A_{1L} = (SIDE - W_{L}) • H_{L} Vertical face T_{1n} = 293.15K

 $A_2 = W_1 \cdot H_2$ Horizontal face $T_{in} = 303.15$ K

 A_{22} = (SIDE - W_{L})• H_{2} Horizontal face T_{M} = 293.15K

Three-Dimensional Sketch of the "Hottest" Area Figure 2.6

TABLE 2
Final Choice for the Dimensions of I.F.V.

HEIGHT FLIGHT	SIDE (m)	PATH (m)	DELTA	PHI
50 (m)	21.88	1119.34	0.56°	2.0°
60 (m)	21.97	1123.89	0.56°	2.5°
70 (m)	22.04	1127.25	0.56°	3.0°
80 (m)	22.09	1129.84	0.56°	3.5°
90 (m)	22.13	1131.96	0.56°	4.0°
100 (m)	22.16	1133.72	0.56°	4.5°

 A_5 = SIDE • H_5 Vertical face T_{in} = 293.15K

 $A_4 = W_4 \cdot H_4$ Vertical face $T_{in} = 303.15K$

At this point we must notice that surface A_5 corresponds to the stack and, since we don't have information about the insulation of this structure, we approximate its temperature with that of the surface with T = 303.15 $^{\circ}$ K

 $A_6 = 4.43 \cdot 10^4 \text{ cm}^2$ Horizontal face $T_{in} = 425.4\text{K}$

 $A_{\tau} = SIDE \cdot H_{\tau} - (\pi/4) \cdot W_{t_{i}}^{2}$ Horizontal face $T_{in} = 293.15K$ For the order of magnitude of PATH i.e. the distance between missile and ship we will approximate the thermal emission of each individual surface as point source emission. These "point sources" will be:

 $(A_1 + A_{11}), (A_2 + A_{21}), A_3, A_4, A_5, A_6, A_7.$

with corresponding mean altitudes from sea level:

 $H_{ML} = H_{MII} = H_{L} / 2$

 $H_{\mu \lambda} = H_{\mu \lambda} = H_{\lambda}$

 $H_{M3} = H_1 + H_3 / 2$

 $H_{M4} = H_1 + H_3 + H_4 / 2$

 $H_{Mg} = H_{L} + H_{g} + H_{4} + H_{5} / 2$

Hmg= H1 + H3 + H4 + H5

H_{M3}= H₁ + H₃

The data for the above calculations are:

 $W_{L} = 1,620 \, \text{cm}$

 $H_1 = 720 \text{ cm}$

 $H_{3} = 210 \, \text{cm}$

 $H_3 = 900 \, \text{cm}$

 $H_{1} = 1,260 \text{ cm}$

 $W_4 = 900 \, \text{cm}$

 $H_4 = 360 \, \text{cm}$

 $H_5 = 180 cm$

III. ATMOSPHERIC TRANSMITTANCE USING LOWTRAN 6

A. DESCRIPTION OF LOWTRAN 6

LOWTRAN 6 is the latest issue of LOWTRAN code that has been designed to calculate the transmittance of the atmosphere over spectral bands from 0.25µm to 28.5µm along paths of various inclinations and, of course, through different atmospheric profiles and parameters.

In order to use LOWTRAN 6 program we have to fill four (4) cards (groups) of input data.

Cards 1 and 2 require the significant meteorological parameters for the atmospheric profile. Card 3 requires the geometry of the problem i.e. initial and final altitude and the path length. Card 4 is related to the wavelength range and limits.

B. DATA USED

For card 1 we used data for the Mediterranean Sea and more specifically for the Aegean Sea area and the region from Crete to North Africa.

For this area we have two different atmospheric profiles, namely Midlatitude Summer for months October through April, and Tropical for months May through September [Ref. 6].

For card 2 we used again data from [Ref. 6] for each month individually and for card 3 we used all the possible combinations between final altitude and PATH from Table 2 and for the initial altitude the mean altitudes of each "point source" of the "hottest" ship area as given in Section G of Chapter II.

At this point we have to choose one more parameter that will be needed for our problem : the wavelength range of the detector operation. We decide that this range will be, in first approximation, in the 5um window (i.e. $\lambda_1=4.20\mu m$ - $\lambda_2=5.18\mu m$) [Ref. 4:ch 5, p.89].

Lowtran 6 has the advantange of giving us, besides the average transmittance between two limits, the spectral transmittance as well. From this spectral distribution we can reject certain values of very low spectral transmittance and therefore the wavelength range can be narrower. This peculiarity will have an influence on card 4.

C. RESULTS

Running LOWTRAN 6 for all the flight altitudes of the missile and for both Midlatitude and Tropical atmospheric profiles, we observed that, for wavelengths less than 4.435µm and greater than 5.0µm, the spectral transmittance generally shows values less than about 0.10. Deciding to reject these low values we choose as the final spectral range of the detector operation to be from 4.435µm to 5.0µm

In Figure 3.1 you will find the spectral transmittance in the above range for both atmospheric profiles.

Using the final wavelength range we calculated the average transmittance for all the possible combinations between atmospheric profile, missile flight altitude and individual "point sources" of the ship's tracked area. Tables 23 to 46 in Appendix C demonstrate both the data and total average transmittance for each month.

From the data of the above tables, it is obvious that the influences of both missile flight altitude (from 50m to 100m) and the relative differences between the individual "point source" altitudes (4m to 16m) are very weak on the

average transmittance. Averaging in both directions in this two dimensional table results in a unique representative value for each month. Checking these average values we observe again a relatively small difference between months belonging in the same atmospheric profile. Since a unique value of average transmittance of each atmospheric profile would be very convenient for the calculations to follow, we average over several months yielding the following final values:

Midlatitude summer : 0.4711

Tropical: 0.4259

Spectral Transmittance in the Window of 4.435µm-5.00µm Through LOWTRAN 6 Figure 3.1

IV. CALCULATION OF SHIP AND BACKGROUND RADIANCE

A. NORMAL COMPONENT OF SHIP RADIANCE DUE TO THERMAL RADIATION

. The purpose of this calculation is to find, by integrating the Plank's radiation law, the radiance in the given wavelength range.

From Plank's radiation law we know that:

$$M_{\lambda}(T) = [2\pi c^{2}h/\lambda^{5}] \cdot [1/(e^{\frac{hc}{k\lambda T}} - 1)]$$
 (W/cm² cm)
or
 $M(T) = C_{i}\lambda^{5}/(e^{\frac{C_{\lambda}}{\lambda T}} - 1)$

where:

$$c_{\perp} = 2\pi c^{2} h = 3.7415 \cdot 10^{12} \quad (W \cdot cm^{2})$$
 $c_{\perp} = hc/k = 1.4388 \quad (cm \cdot K)$

The above formula is in the form of:

$$M(T)=C \cdot x^{m}(e^{x}-1)$$

Integrating this between the band limits λ_1 , λ_2 , we will find the in-band energy radiance [Ref. 4:p. 19]. Therefore:

$$L_{\Delta\lambda} = \mathcal{E}_0/\pi \int_{\lambda_{\perp}}^{\lambda_{\lambda}} (T) d\lambda = \left[C_1 \mathcal{E}_0/\pi C_{\lambda}^4 \right] \cdot T^4 \cdot \sum_{\perp} (W/cm^2 \cdot Sr)$$

Where:

$$\sum_{i} = |\sum_{m=1}^{\infty} m^{4} [(m \cdot x)^{3} + 3(m \cdot x)^{2} + 6(m \cdot x)^{2} + 6m \cdot x + 6] e^{-w_{i}x}|_{x_{i}}^{x_{2}}$$

and $x=C_2/\lambda T$

$$\sum_{i} = \sum_{m>i} \tilde{m}^{wx_{i}} \tilde{m}^{-4} [(m \cdot x)^{3} + 3(m \cdot x)^{2} + 6m \cdot x + 6] - e^{wx_{i}} \tilde{m}^{-4} [(m \cdot x)^{3} + 3(m \cdot x)^{2} + 6m \cdot x + 6]$$

In the above relations, the temperature T is taken from Tables 5, 18 in Appendix A.

B. NORMAL COMPONENT OF SHIP RADIANCE DUE TO REFLECTED SOLAR ENERGY

We have already calculated the normal component of solar irradiance E incident on each ship surface. In the energy balance calculation we considered that the solar energy absorbed by the surface is $E_{s\eta}$ a. Obviously the amount $(1-a_{\circ}) \cdot E_{s\eta}$ is reflected.

Defining as $D(0-\lambda)$ the percentage of the solar constant associated with wavelengths shorter than λ we take [Ref. 4:p. 17].

Therefore

$$\Delta D = D(0-5.000) - D(0-4.435) = 0.2039546\%$$

In conclusion we have that the normal reflected solar energy per m² from $\lambda_1=4.435\mu m$ to $\lambda_2=5.00\mu m$ is:

$$(1-a_o) \cdot \Delta D \cdot E_{sn} = (1-a_o) \cdot 2.04 \cdot 10^{-3} \cdot E_{sn} \quad (W/m^2)$$

In order to determine the normal component of radiance due to reflection we must divide the above value by π and in order to transform into cm² we must divide by 10⁴. Finally the normal component of radiance due to reflection in $(W/cm^2 \cdot Sr)$ from $\lambda_1=4.435 \mu m$ to $\lambda_2=5.00 \mu m$ will be: $L_{RA\lambda}=(1-a_0)\cdot 2.04\cdot 10^{-3}\cdot E_{sn}/\pi \cdot 10^4=(1-a_0)\cdot 4.08\cdot 10^{-8}\cdot E_{sn}$ $(W/cm^2 \cdot Sr)$

C. NORMAL COMPONENT OF STACK EXIT PLANE RADIANCE

So far we know that the stack exit plane area is $4.43 \cdot 10^4$ cm² and that the corresponding uniform temperature is 425.4K.

For simplicity we consider that the greater part of the exaust gas is $\mathrm{CO_2}$. Since $\mathrm{CO_2}$ is a selective emitter we need to carry out the approximation shown in Figure 4.1 which partially accounts for atmospheric absorption and approximates the peak with a constant emissivity over a limited

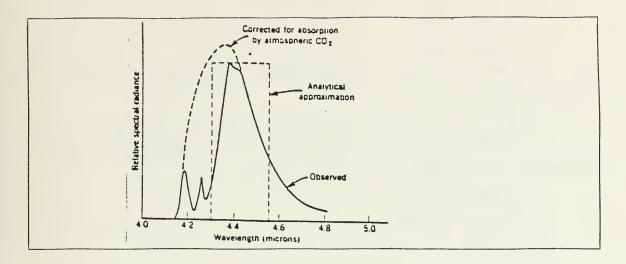


Figure 4.1 The 4.4um Emission Band of CO₂ [Ref. 12]

range. Taking the band to be from $\lambda_1=4.35$ um to $\lambda_2=4.55$ um and the emissivity 0.5 we find for the radiance:

$$L = e_0 \int_{4.35}^{4.55} d\lambda = (\epsilon_0 \sigma T^4 / \pi) \int_{4.35}^{4.55} M_{\lambda} d\lambda / \int_{0}^{\infty} M_{\lambda} d\lambda = (\epsilon_0 \sigma T^4 / \pi) \cdot (0.05)$$

$$= (0.5) \cdot (5.67 \cdot 10^{-12}) \cdot (425.4)^4 \cdot (0.05) / \pi$$

$$= 1.478 \cdot 10^{-3} \quad (W / cm^2 \cdot Sr)$$

Since our band of interest overlaps with half of the $4.4\mu m$ emission band of CO_2 the above value must be divided by 2. Therefore:

$$L=7.39 \cdot 10^{-4}$$
 (W/cm² · Sr)

D. TOTAL SHIP RADIANT INTENSITY IN THE DIRECTION FROM SHIP TO MISSILE

As mentioned in Section G of Chapter II, we will consider the surfaces $(A_1 + A_{11})$, $(A_2 + A_{21})$, A_3 , A_4 , A_5 , A_6 , A_7 , as "point sources". For each source we know the corresponding radiance due to thermal radiation and reflection i.e.

 $L_{71} = L_{\Delta\lambda_1} + L_{R\Delta\lambda_1}$ $L_{711} = L_{\Delta\lambda_1} + L_{R\Delta\lambda_1}$ $L_{72} = L_{\Delta\lambda_2} + L_{R\Delta\lambda_2}$ $L_{72} = L_{\Delta\lambda_2} + L_{R\Delta\lambda_2}$ $L_{73} = L_{\Delta\lambda_3} + L_{R\Delta\lambda_3}$

 $L_{74} = L_{624} + L_{R624}$ $L_{75} = L_{625} + L_{R625}$ $L_{77} = L_{627} + L_{R625}$

For L_{TG} we have the unique value given in section C of chapter III

In order to obtain the normal radiant intensity (W/Sr) of each surface we must multiply the corresponding values of each area and radiance. Doing this we take:

 $I_{1M} = L_{T1} \cdot A_{1} + L_{TM} \cdot A_{11}$ $I_{2M} = L_{T2} \cdot A_{2} + L_{T22} \cdot A_{22}$ $I_{8M} = L_{78} \cdot A_{3}$ $I_{4M} = L_{74} \cdot A_{4}$ $I_{5M} = L_{75} \cdot A_{5}$ $I_{6M} = L_{76} \cdot A_{6}$ $I_{7M} = L_{71} \cdot A_{7}$

The important magnitude for us is the radiant intensity in the direction of source - missile.

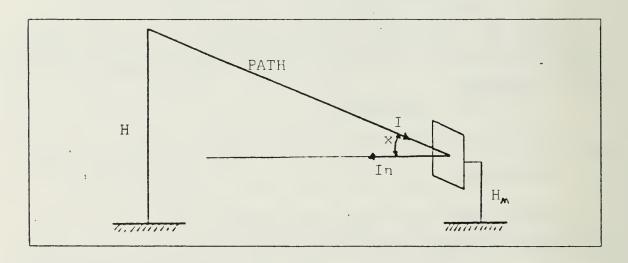


Figure 4.2 Relative Position Between Surface- Missile

In Figure 4.2 you will find a sketch representing the relative position between the surface and the missile. Based on that we conclude this the radiant intensity will be:

 $I=I_N \cdot cosx$ for a vertical face

I=I_N•sinx for a horizontal face where x=sin¹(H-H_M/PATH) The resulting relations will be:

I_ = I H • COSX

 $I_1 = I_{2N} \cdot sinx$

 $I_b = I_{3h} \cdot cosx$

 $I_4 = I_{4N} \cdot \cos x$

 $I_5 = I_{5n} \cdot cosx$

 $I_6 = I_{6N} \cdot \sin x$

 $I_1 = I_N \cdot sinx$

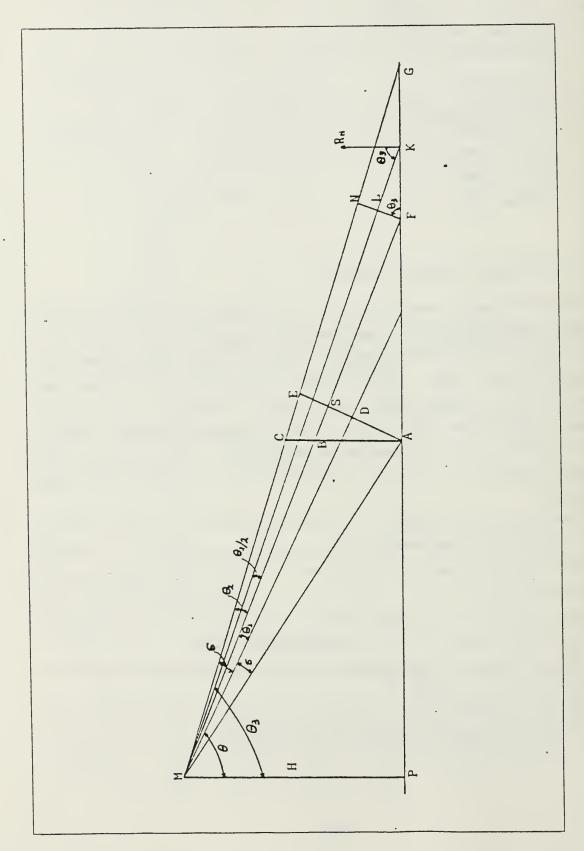
E. BACKGROUND RADIANT INTENSITY

In the ideal case in which the detector tracks perfectly on the ships "hottest" area and the square I.F.V. will fit perfectly on that the detector will still see some areas outside of the ship. This is due to the fact that the upper construction of the ship has not a perfect rectangular shape. Therefore the detector will receive signal from an area of the sea that radiates behind the ship. Since this signal constitutes noise for the detector, it has important meaning.

In Figure 4.3 you will find the geometry from which we will determine the background radiant intensity. We must notice that the sketch in Figure 4.3 has been magnified for convenience in the ship region. In other words angle $(\Theta+\delta)$ is very close to 90° and the side of I.F.V. AE is ASE is practically equal to the ship dimension ABC.

AB represents the height of surface A_{x} [Figure 2.6] from the sea level or $AB=H_{1}+H_{2}$

BC then represents the path through which the detector will see the sea area FG past the stack.



Geometry for Determination of Background Radiant Intensity Figure 4.3

The width of this area is (SIDE- W_1) which represents the free horizontal dimension over the surface A_2 .

From the geometry of Figure 4.3 we have:

DS=AS-SIDE

 $\Theta_{1} = tan^{-1} (DS/PATH)$

02 = 5 - OL

MA=PATH/cos8

 $\Theta = \cos^{-1}(H/MA)$

 $PF=H \cdot tan(\Theta + \delta + \Theta_{\lambda})$

PG=H•tan(Θ+28)

FG=PG-PF

03 = 0 + 5 + 01 + 01 / 2

MK=H•tan03

AREA=FG • [SIDE-W6] • cosO8

Table 3 demonstrates the result of the above calculations for six different flight altitudes.

TABLE 3
Geometrical Parameters for Observed Sea Area

H (m)	FG (m)	- 0 (deg)	MK (m)	AREA (m²)
50	333.67	88.415	1806.98	144.90
60	230.54	87.915	1648.07	131.68
70	180.09	87.410	1547.47	127.76
80	144.50	86.91	1481.94	122.29
90	116.07	86.4	1430.50	114.42
100	99.42	85.9	1395.07	111.50

For the sea water we know that the average temperature in the geographical area of our interest is 13°C=286.15K for Midlatitude Summer and 18°C=291.15Kfor Tropical atmosphere [Ref. 6].

For the sea water emissivity we chose the value of 0.96 [Ref. 10:p.07].

For the calculation of the normal component of the sea radiance (W/cm²•Sr) in the range λ_1 =4.435 μ m to λ_2 =5.00 μ m we will use again the formulas from section A of chapter IV. This value must be multiplied by the water emissivity =0.96. For convenience we consider the observed area of sea as a point source.

The normal component of the radiant intensity of this point source it will be:

Iwn=Lnas AREA

And in the direction leading to the missile will be: $I_{\mathbf{w}} = I_{\mathbf{w}, \mathbf{v}} \cdot \cos \theta$

V. DETECTOR MODELING AND CALCULATION OF S/N RATIO

A. DETECTOR SYNTHESIS

We already studied the geometry of the detection problem which shows us what the detector is required to do, and of course from this side we are interested in the most realistic performance of the detector. Since we do not have available experimental data that could show us the performance of the detector element we have to choose this according to the literature.

From Figure 5.1 [Ref. 10] the possible choices for the detector include photovoltaic or photocontactive Indium Antimonide and Lead Selenide. The detailed data sheets for these detectors [Ref. 12:pp.365-368] show the relative advantages and disadvantages of each. When they are compared on the basis of D*, photovoltaic Indium Antimonide is clearly the best detector. A possible difficulty with this conclusion arises when the responsivities are compared and it is noted that value of Lead Selenide is 10 times higher than it is for the other two types of detector. In an ideal operating environment, that is, in which there are no extraneous sources of noise, the designer is interested in the signal - to - noise ratio of the detector, as described by D* and not the signal level, as described by responsivity. advance we know that during the experimental trials photovoltaic Indium Antimonide had the better performance. Therefore we are led to the decision to use one element of photovoltaic Indium Antimonide, and we can see the detailed data for this material in Figure 5.2 [Ref. 12:p.368].

This one element will be placed in the middle of an assumed two mirror system as in Figure 5.3, with

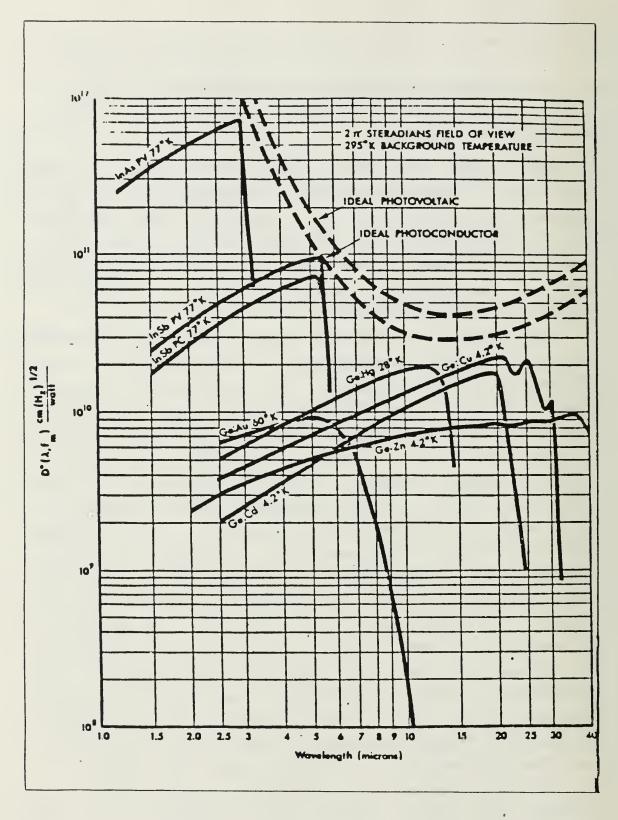


Figure 5.1 The D* of the Detectors in Each Atmospheric Window at the Most Common Operation Temperature [Ref. 10]

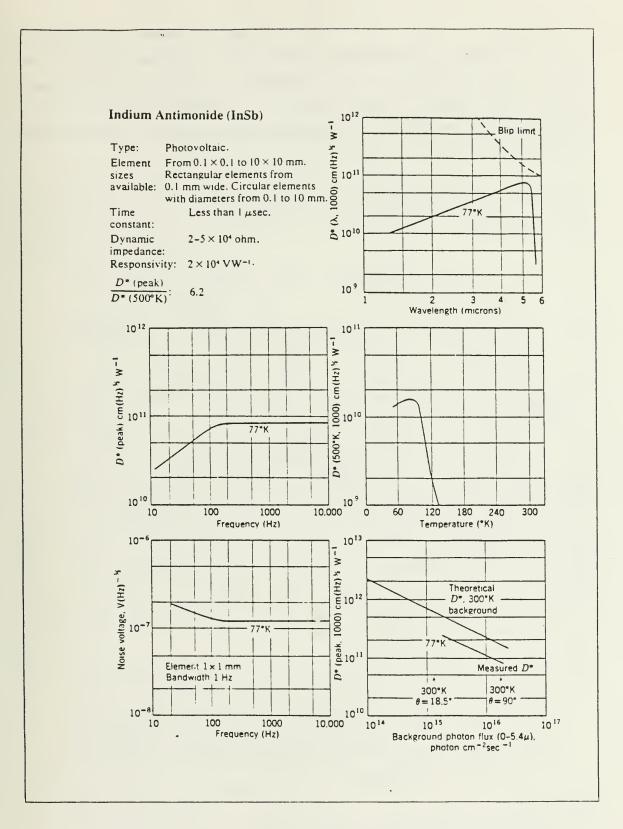


Figure 5.2 Data Sheet for Photovoltaic Indium Antimonide Detector [Ref. 12]

reflectivity 0.9 each mirror and total of two r_0 = 0.9•0.9=0.81. The detector will be considered to be used with a two-mirror system designed so that the detector element will subtend area equal to the I.F.V. at a distance of PATH. Thus we have to determine the size of the sensitive element (detector).

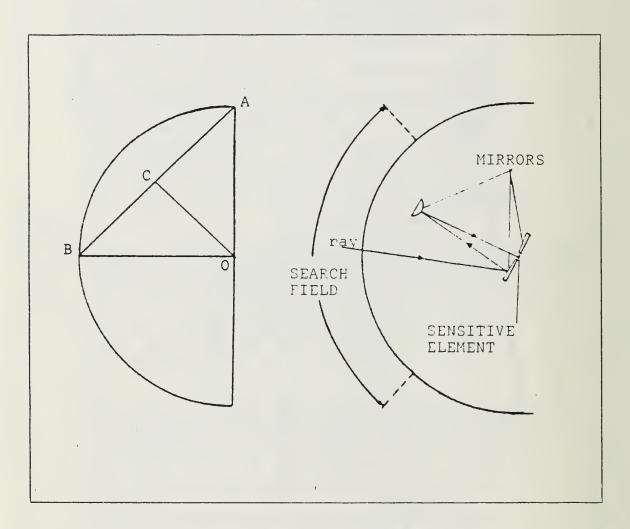


Figure 5.3 System of Two Mirrors in the Receiver

We know that the area of the detector A is given by: $A_d = w \cdot f^2$

where

w=Instantaneous Field of View in (Sr)
w=(SIDE•SIDE)/(PATH)²=(2200•2200)/(PATH)² (Sr)

f=equivelant focal length

and for PATH we have to use the values from Table 2 which is function of the flight altitude.

For the our purpose of work a very good and accurate approach to calculate the equivalent focal length of the system of two mirrors is the following.

First we have to choose the diameter of the entrance aperture D_{\bullet} =12cm. This dimension is good enough to have an aerodynamic shape in the nose of the missile. Thus from the Figure 5.3 we have:

$$(AB)^2 = (AO)^2 + (OB)^2 = 72$$
 and $AB = 8.4$ cm

$$(AO)^2 = (AC)^2 + (CO)^2$$
 or

$$(CO)^2 = (AO)^2 - (AC)^2 = (AO)^2 - (AB/2)^2 =$$

= $(CO)^2 = 6^2 - (8.49/2)^2 = 17.98 \text{ cm}^2$

and CO=4.24cm

We can define the distance CO as the equivalent focal length f=4.24cm

And

$$A_d = w \cdot f^2 = (2200)^2 \cdot (4.24)^2 / (PATH)^2$$

Substituting the values of PATH for each flight altitude gives A and the square root of this gives us the side of the detector element. Table 4 demonstrates the dimensions of the detector element shows that with reasonable detector size the I.F.V. at about 1125 meters slant path range will include the hot region of the ship regardless of the flight altitude.

B. SIGNAL VOLTAGE

The general formula that gives the signal voltage is: $V_5 = [A_0 \cdot r_0 \cdot R/(PATH)^2] \cdot [\int_{\lambda}^{\lambda_1} I_{\lambda} I_{\alpha}(x) dx]$

Where

Ao: Area of entrance aperture

 $A_{\bullet} = \pi D^2 / 4 = 113.1 \text{cm}^2$

ro: Optics reflectance

TABLE 4
Dimensions of the Detector Element

FLIGHT ALTITUDE (m)	PATH (m)	DIMENSION OF DETECTOR ELEMENT (wm)
50	1,120	0.83 × 0.83
60	1,124	0.83 x 0.83
70	1,127	0.83 x 0.83
80	1,130	0,83 x 0,83
90	1,132	0.83 x 0.83
100	1,134	0.83 x 0.83

R :responsivity=2.104 (V/W)

 I_{λ} : Spectral radiant intensity of the source

La:Spectral atmospheric transmittance

The above relation can be further simplified using instead of the spectral transmittance an average one as we determined in Section C of Chapter III i.e. 0.4711 for Midlatitude summer and 0.4259 for Tropical atmosphere.

Taking $\mathcal{T}_{\alpha}(\lambda)$ outside of the integral then the quantity $\int_{\lambda}^{\lambda_1} \mathrm{d}\lambda$ becomes simply the inband radiant intensity of the source which has been calculated in section D of chapter III. Hence the signal voltage formula becomes:

$$Vs = A_{\bullet} r_{\bullet} R \cdot C_{\bullet} I_{A \Sigma} / (PATH)^{2}$$

In order to determine the signal voltage due to all the available point sources inside the I.F.V. we use the superposition principle:

$$Vs = [A_{\bullet} r_{\bullet} R_{\bullet} T_{q} / (PATH)^{2}] \cdot [I_{h} + I_{2} + I_{3} + I_{4} + I_{5} + I_{6} + I_{7}]$$

C. NOISE VOLTAGE

The noise voltage is due to two main factors, namely the background noise that is created from the background radiant intensity and the noise from the optical system.

In Section E of Chapter III we determined the radiant intensity of the sea region that the detector observes and we named $I_{\mathbf{w}}$ (W/Sr). The noise voltage of this reason will be:

$$V_{NB} = A_{\bullet} r_{\bullet} R_{\bullet} T_{q} I_{W} / (MK)^{2}$$

where MK is determined in Figure 4.3 and indicates the average path between the observed sea surface and the detector.

The noise signal due to the optical system is given by: [Ref. 12].

$$V_{NO} = R \cdot (A_{\bullet} \Delta f)^{1/2} / D^{*}$$

where

D*:Detectivity with average value $9 \cdot 10^{11}$ [cm(Hz) $^{1/2}$ /W] in the wavelength range $\lambda = 4.435 \mu m$ to $\lambda = 5.00 \mu m$

A_d:Area of the detector as determined in previous section of the present chapter.

R : Responsivity

Δf :equivalent noise bandwidth

The equivalent noise bandwidth is given by

where

Td: dwell time (time required for the image of the target to pass across the detector).

where

c:number of detector elements (1 for our case)
.

Q:Time rate of search.

and

where

Q:size of search field in (Sr)

G:frame time (time required to scan the entire search field). For our case we chose 10msec.

Therefore

$$\Delta f = 1/2 I_0 = \dot{\Omega}/2 \cdot w \cdot c = \Omega/2 \cdot w \cdot c \cdot I_F$$

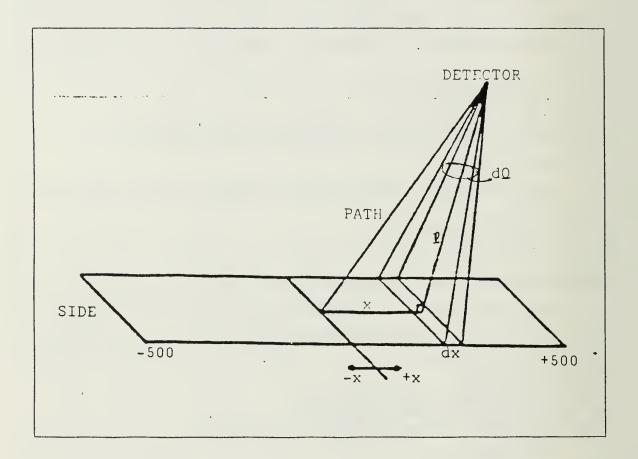


Figure 5.4 Total Detector Search Field

In Figure 5.4 you will find the geometrical representation of the total search field which will have total length of 1000m and is scanned by the detector in =10msec.

The total solid angle that the search field subtends at the detector is:

Q= dQ $dQ=dx \cdot SIDE/\mathcal{Q}^2$ $\mathcal{L}^2 = (PATH)^2 - x^2$

$$dQ = [SIDE/(PATH)^2 - x^2] \cdot dx$$

$$Q = (SIDE) dx/(PATH)^2 - x^2$$

Since (PATH)² >500² always, from the tables we get:

Therefore

$$\Delta f = (1/2(SIDE) \cdot c \cdot \mathcal{T}_{f}) \cdot ln[(PATH+500)/(PATH-500)]$$

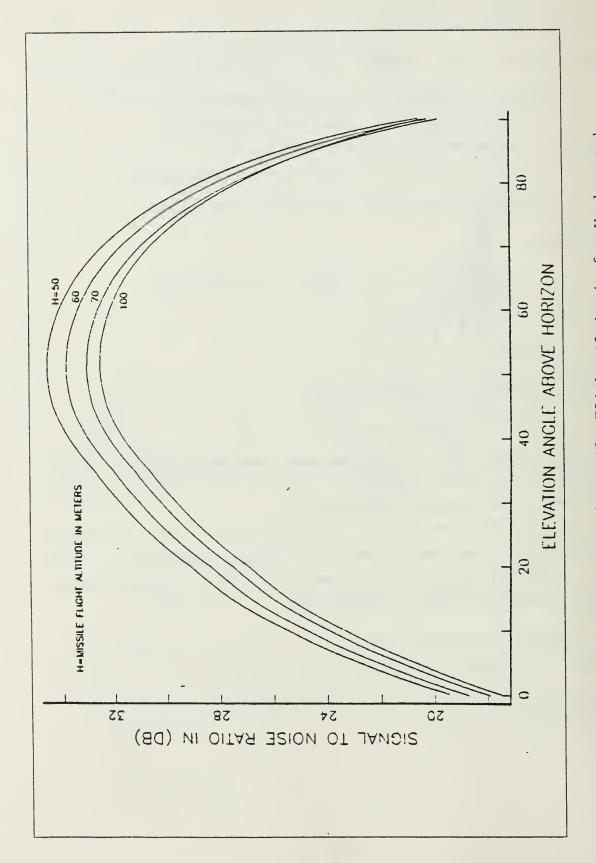
The total noise voltage is

And the signal to noise ratio is:

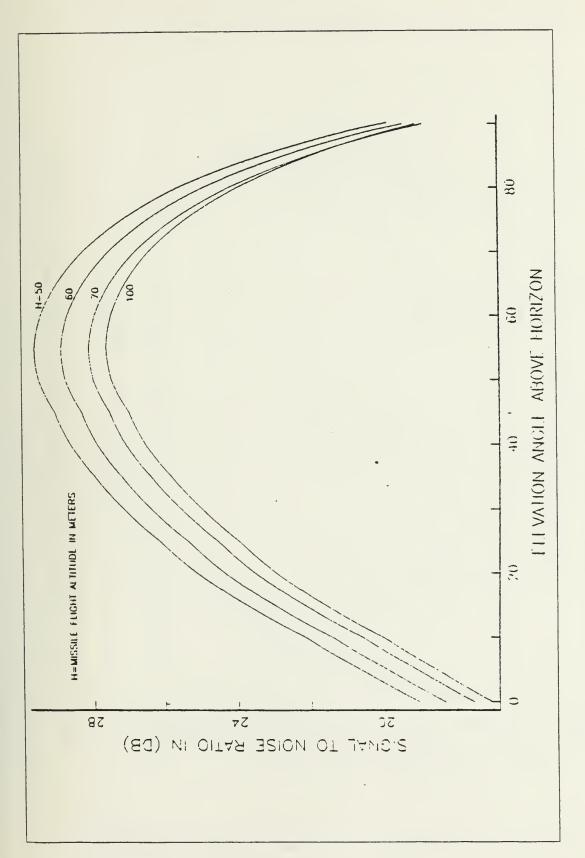
$$Vs/Vn=Vs/(V_{NB}+V_{NO})$$

In figures 5.5 to 5.22 we can see the summary of the results of the above calculation for all the possible combinations of the relative parameters.

In Appendix D we can see the influence of sky condition to S/N ratio for the two extremes flight altitude i.e. 50 (m) - 100 (m) and for all the combinations between atmospheric profile and paint emissivity.



Influence of Missile Flight Altitude for Unobscured Sun, &=0.80, Midlatitude Summer Figure 5.5



Influence of Missile Flight Altitude for Sun with Light Clouds, &=0.80, Midlatitude Summer Figure 5.6

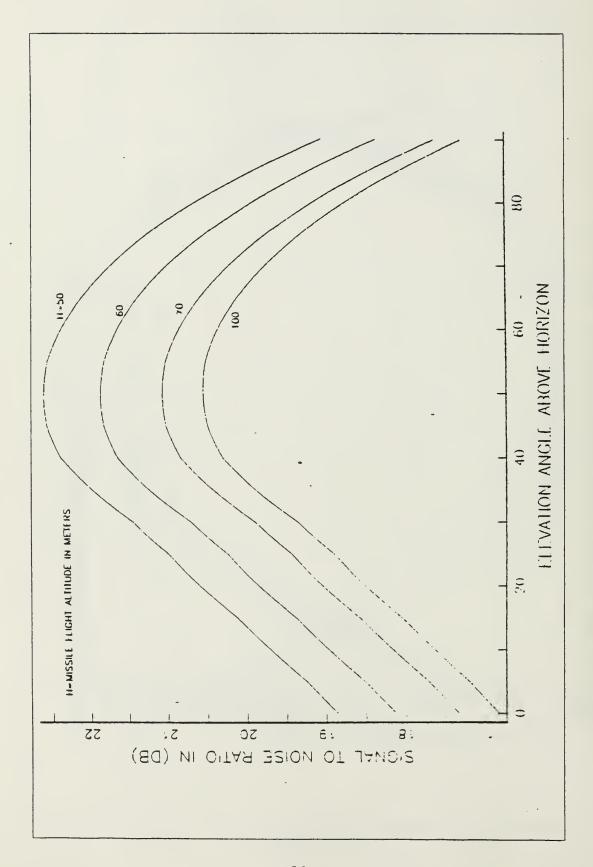


Figure 5.7 Influence of Missile Flight Altitude for Sun with Heavy Storm Clouds, &=0.80, Midlatitude Summer

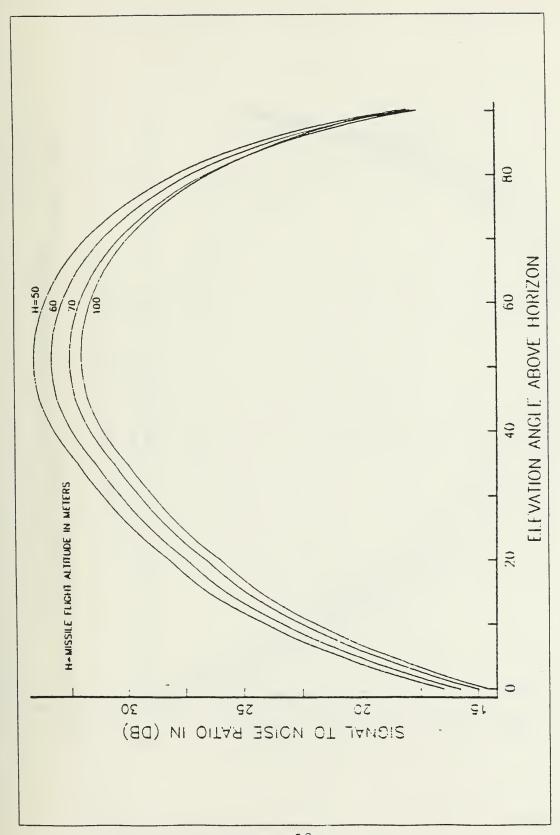
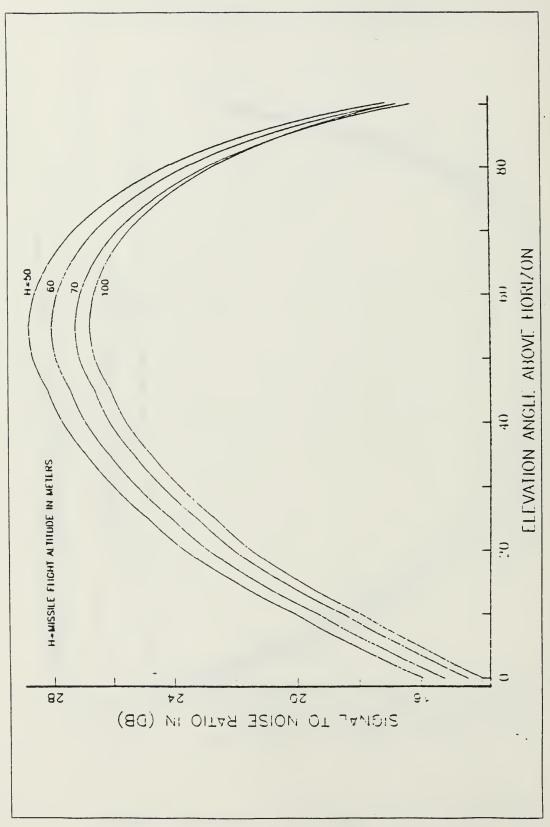
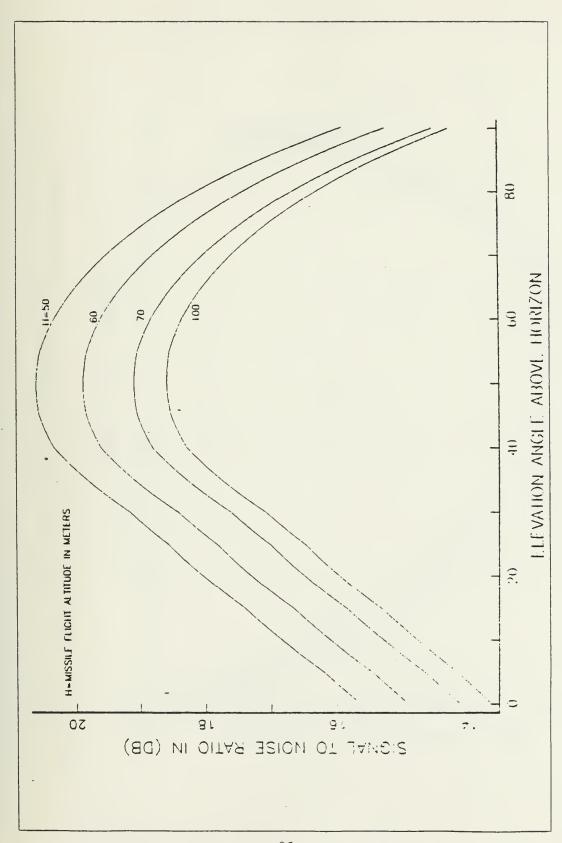


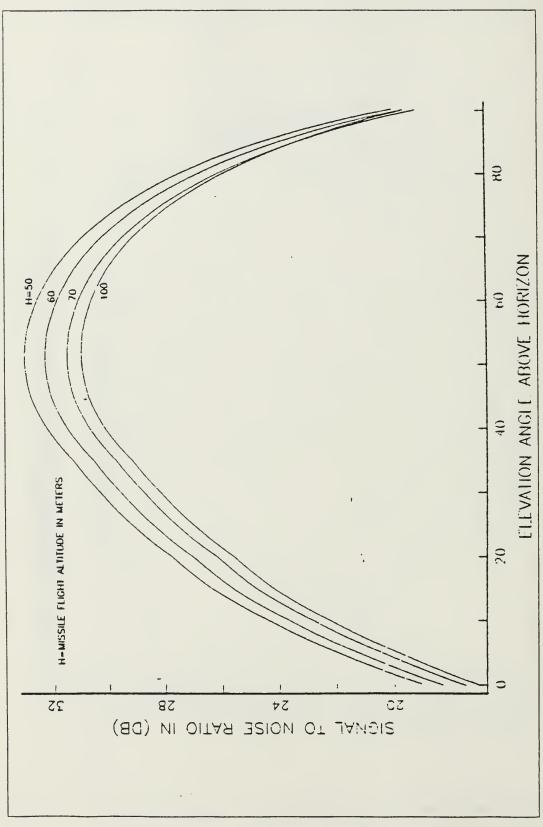
Figure 5.8 Influence of Missile Flight Altitude for Unobscured Sun, &=0.55, Midlatitude Summer



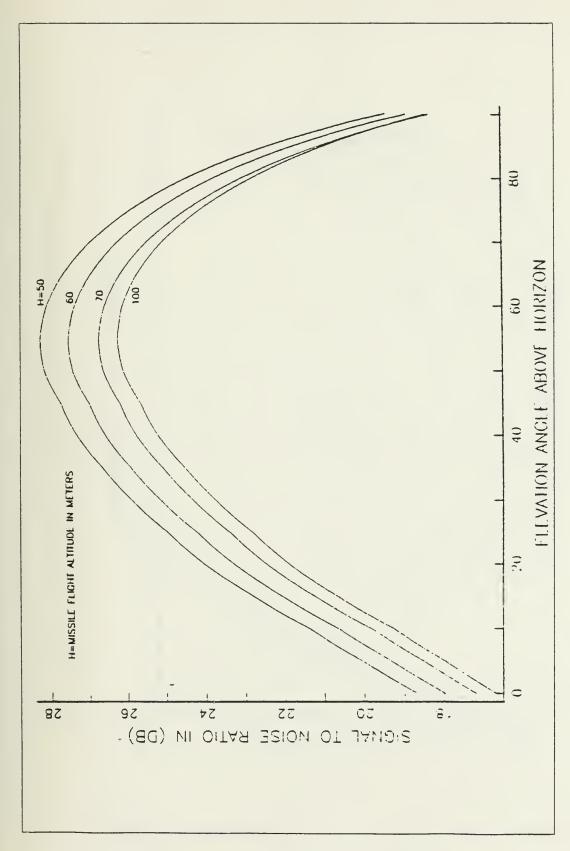
Influence of Missile Flight Altitude for Sun with Light Clouds, &=0.55, Midlatitude Summer Figure 5.9



Influence of Missile Flight Altitude for Sun with Storm Clouds, £e=0.55, Midlatitude Summer Figure 5.10 Heavy



Influence of Missile Flight Altitude for Unobscured Sun, $\mathcal{E}_{\!\boldsymbol{c}} = 0.80$, Tropical Atmosphere Figure 5.11



Influence of Missile Flight Altitude for Sun with Light Clouds, £=0.80, Tropical Atmosphere Figure 5.12

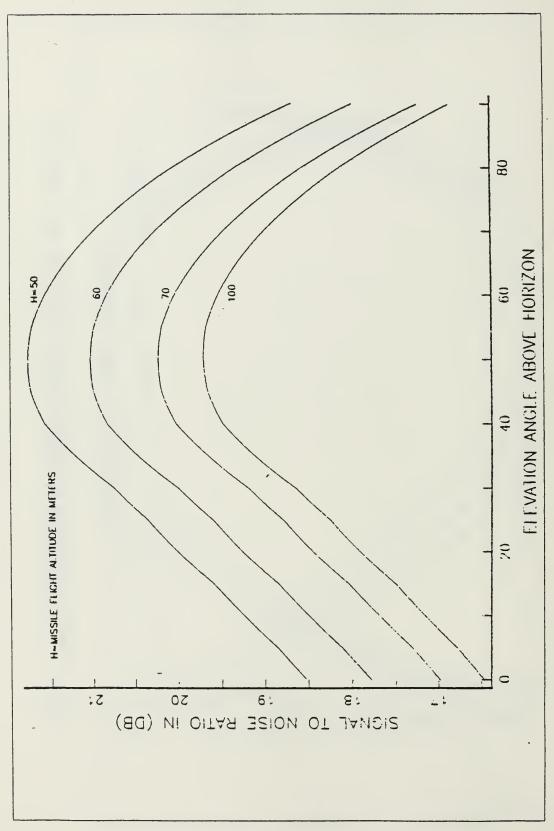


Figure 5.13 Influence of Missile Flight Altitude for Sun with Heavy Storm Clouds, &=0.80, Tropical Atmosphere

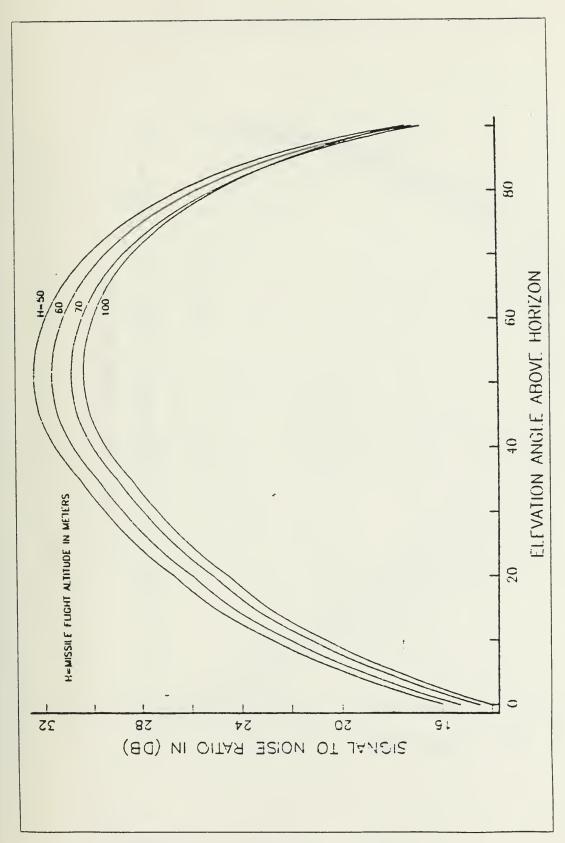
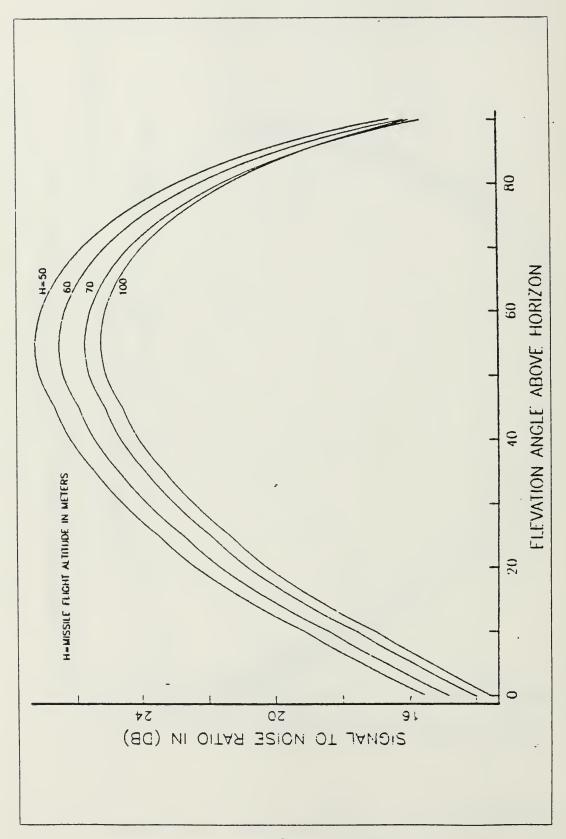
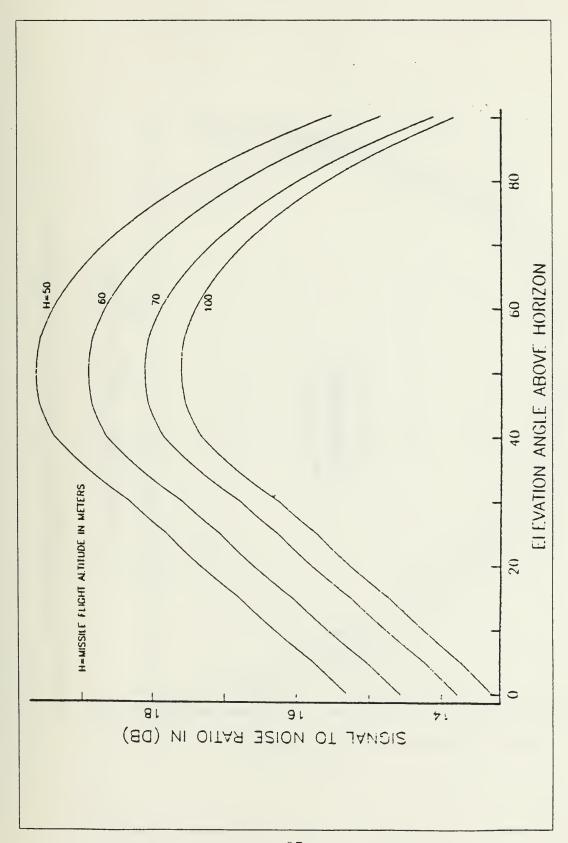


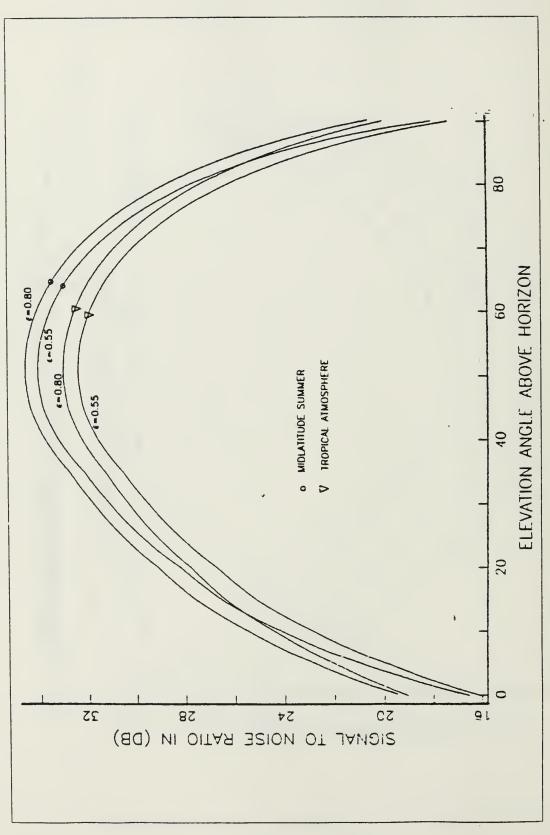
Figure 5.14 Influence of Missile Flight Altitude for Unobscured Sun, &=0.55, Tropical Atmosphere



Altitude for Sun with Atmosphere Influence of Missile Flight Light Clouds, £=0.55, Tropical Figure 5.15



Influence of Missile Flight Altitude for Sun with Storm Clouds, £.=0.55, Tropical Atmosphere Figure 5.16 Heavy



7 Influence of Emissivity and Atmospheric Profile for Flight Altitude 50m and Unbscured Sun Figure 5.17

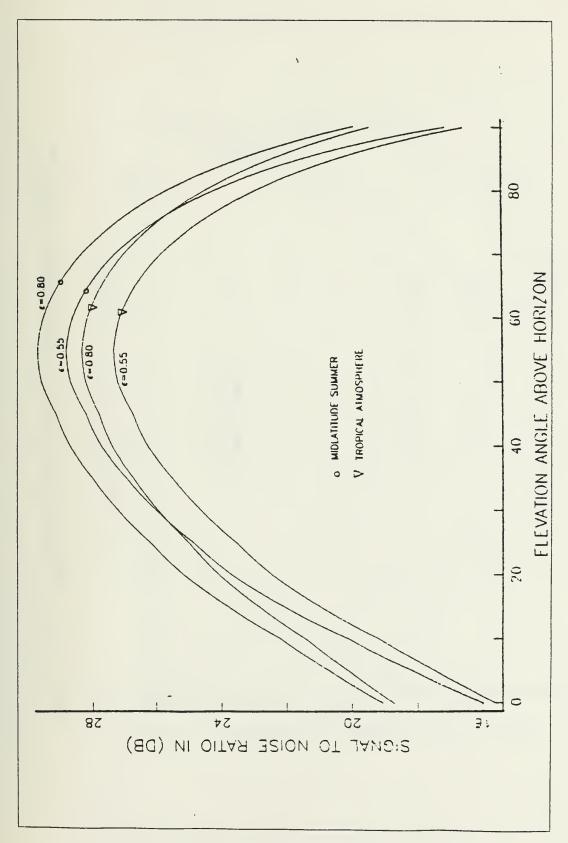


Figure 5.18 Influence of Emissivity and Atmospheric Profile for Flight Altitude 50m and Sun with Light Clouds

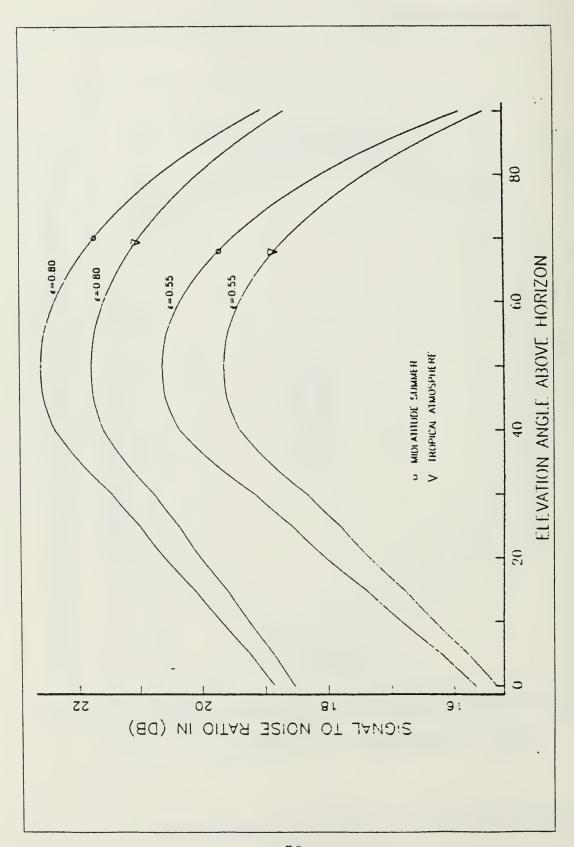


Figure 5.19 Influence of Emissivity and Atmospheric Profile for Flight Altitude 50m and Sun with Heavy Storm Couds

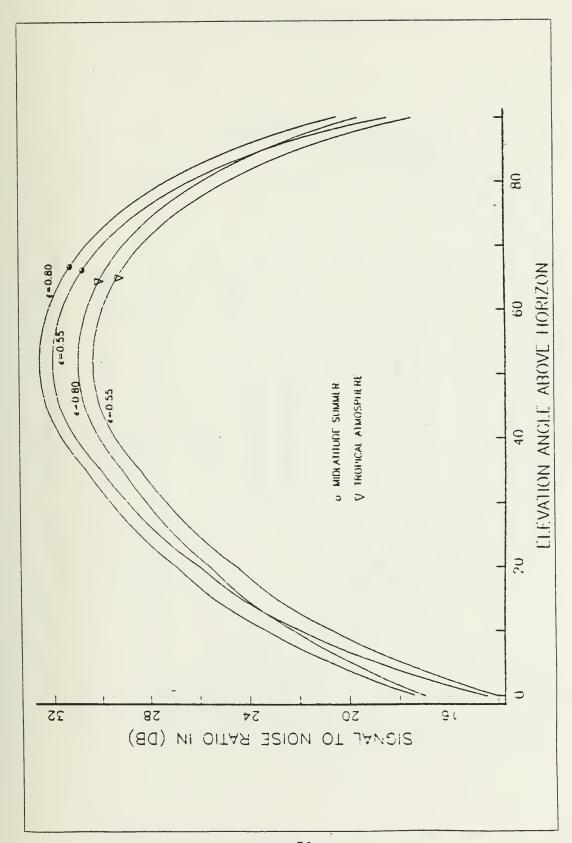


Figure 5.20 Influence of Emissivity and Atmospheric Profile for Flight Altitude 100m and Unbscured Sun

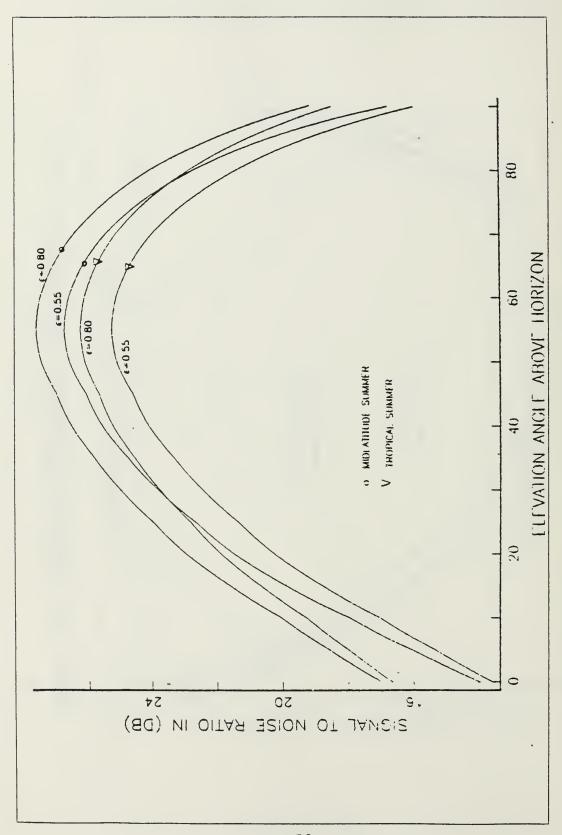
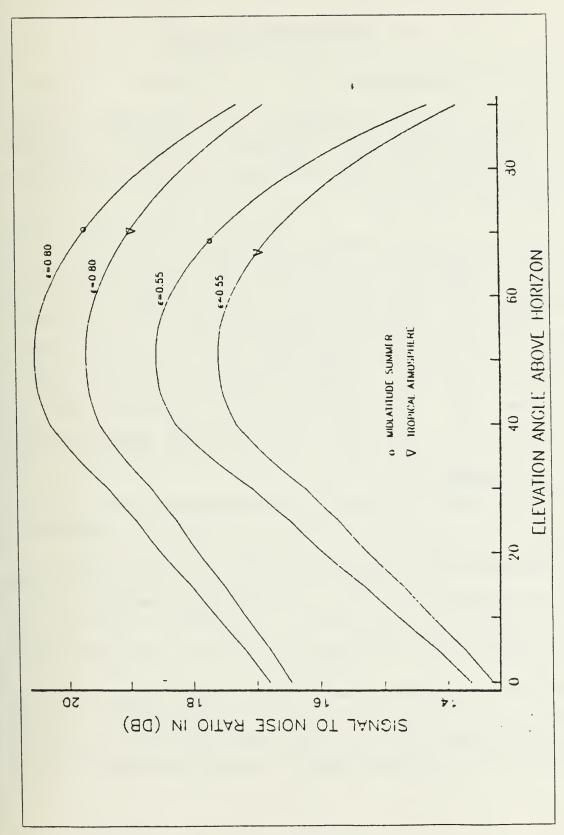


Figure 5.21 Influence of Emissivity and Atmospheric Profile for Flight Altitude 100m and Sun with Light Clouds



Influence of Emissivity and Atmospheric Profile Altitude 100m and Sun with Heavy Storm Couds Figure 5.22 for Flight

VI. RESULTS AND DISCUSSION

A. SHIP BODY TEMPERATURE

Tables 5 to 18 of Appendix A show the effects of the ship internal temperature, atmospheric profile, ship body paint emissivity, sun elevation angle and sky condition on the ship body temperature.

For both horizontal and vertical faces higher paint emissivity and ship internal temperature result in higher ship body temperature. Higher internal temperature corresponds to less ship body heat losses towards the internal compartments. Moreover higher paint emissivity (absorptivity for gray paint) corresponds in higher solar energy absorption and, for the 300K ship body temperature level, the dominant mechanism of heat transfer is that of convection. Hence the increase of the paint emissivity does not correspond to a significant increase of the energy radiated by the ship body, and the net result is an increase of the ship body temperature.

The sky condition appears to be significant as well. The more clear it is the more solar energy there is incident on the ship, resulting in an increase of its body temperature.

As far as the atmospheric profile is concerned, the Tropical Atmosphere results in higher body temperature than that of Midlatitude Summer since the ambient temperature is higher and the ship body heat losses towards the environment are less.

The sun elevation angle above the horizon affects vertical and horizontal faces in different ways. We know that the incident solar energy at sea level is an increasing function of the elevation angle. For an horizontal face the

temperature appears to be an increasing function of the elevation angle, the sine of which is used for the calculation of the incident angle effect. For vertical faces the cosine of the elevation angle is used and the temperature appears a maximum in the vicinity of 50°. The explanation is that, up to this value, the increasing effect of the solar energy dominates the decreasing one of the cosine function. After the value of about 50° the first derivative of the solar energy function becomes small and the cosine behavior dominates so that the net result is a decreasing temperature function from 50° to 90°.

As we see in the ship tracked area geometric analysis is constituted from both vertical and horizontal faces, but the vertical ones have much larger projected surface areas. Therefore the radiant energy from the tracked area is due mainly to that of the vertical faces and as we will see from the S/N ratio results they appear maximum at a value of elevation angle close to 50° .

B. DETECTOR S/N RATIO

Figures 5.5 to 5.22 show the effects of the sky condition, sun elevation angle and missile flight altitude on the detector S/N ratio.

In general we observe that the variation of the S/N ratio is similar to that of the ship body temperature. That means that the radiant energy from the ship body due to its temperature is the main thermal source for the detector. At this point, it is useful to discuss the influence of the stack exit plane thermal radiation. From our calculation we know that the radiant intensity of the CO in the stack exit plane has a magnitude of the order of 104 (W/cm²Sr) and that of the ship body of the order of 103 to 102. Moreover the stack exit plane area is much smaller than the total tracked

area and the stack exit radiant flux is significantly smaller than this of the ship body tracked surface.

Sky condition and the paint emissivity affect S/N ratio similarly to the ship body temperature. The more clear is the sky and the higher is the paint emissivity the higher is the S/N ratio.

The atmospheric profile affects S/N in a different way. While the Tropical Atmosphere gives higher ship body temperature, it results in lower S/N ratios comparing with the Midlatitude Summer. The explanation for that is that the increase of the ship body radiance, due to its temperature increase, is less than the background noise increase (higher sea temperature). Moreover the atmospheric transmittance in the 5um window for the Tropical Atmosphere is about 10% less than this for Midlatitude Summer.

The missile flight altitude appears to be important as a parameter. The higher it is the lower is the S/N ratio and this decrease is higher between 50 to 70 (m) than between 70 to 100 (m). The explanation is that although higher missile flight altitude corresponds to lower background area (sea), the path between missile head and ship becomes larger and its square appears in the denominator of the signal voltage formula. Therefore the net result is to give lower values of S/N ratio.

The influence of the sun elevation angle above the horizon is similar to that on the ship body temperature and it has been discussed earlier.

C. RECOMMENDATIONS

Based on the relative influence of each parameter which has been varied in our analysis we recommend:

From the ship design point of view, lower internal temperature or, when it is not feasible, better wall

insulation, especially in the area where significant heat generation exist. In addition to that, paint of low emissivity and absortivity is desirable.

From the missile point of view we recommend lower flight altitude. At this point we must notice that, lowering the flight altitude, we will reach a point where the background area (sea) will be very large, resulting in higher background noise and the coast or even the sun itself could constitute background noise for the detector. Therefore the flight altitude must be determined, in a certain design, accounting for all these conflicting parameters. Another solution could be the use of higher flight altitude, that assures us low background area, and to improve the detector optics design that will balance the loss of the signal voltage due to the greater path length.

$\frac{\texttt{APPENDIX} \ \, \texttt{A}}{\texttt{TABLES} \ \, \texttt{FOR} \ \, \texttt{SHIP} \ \, \texttt{BODY} \ \, \texttt{TEMPERATURE}}$

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TABLE 6

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TABLE 7

Ship Body Temperature for Midlatitude Summer T = 293.15 K and = 0.55

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WITH CLOUDS VERTICAL FACE	96719671867129622964000771867222011 99998888277779666286767778332222211 74447474747474747474747474747 749999999999
SUN LIGHT HORIZONTAL FACE	**************************************
RED SUN VERTICAL FACE	************************************
UNOBSCURED HORIZONTAL VER FACE	278047159470618851885239642198887766 37804716665851885239664219111111111111111111111111111111111
ELEVATION ANGLE	. 000000000000000000000000000000000000

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TABLE 8

VERTICAL FACE SUN WITH HEAVY STORM CLOUDS Summer HORIZONTAL FACE e for Midlatitude 5K and =0.55 VERTICAL FACE WITH CLOUDS SUN HORIZONTAL FACE Temperature T=303.1 するちょうとしゅり ちきてのできる 移と りられき できてし つのの ららら Body VERTICAL FACE SUN ϕ Ship UNOBSCURED HORIZONTAL FACE $\frac{1}{2} \frac{1}{2} \frac{1}$ annananananananananananananananananan ELEVATION ANGLE をできるいの名とのでもとうの名とのできとうの名とのられをとての **エエエエニエエエスところころころころろろろろろろろろろろ**

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TABLE 9

Ship Body Temperature for Tropical T = 293.15K and = 0.80

SUN WITH HEAVY STORM CLOUDS	VERTICAL FACE	
	HORIZONTAL FACE	
WITH	VERTICAL FACE	17407m962m840628495H78267964196 655544mmm22221100696969696969696969696969696969696969
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	HORIZONTAL FACE	Managamananananananananananananananananan
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TABLE 10

Ship Body Temperature for Tropical T = 303.15K and =0.80

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TABLE 11

Ship Body Temperature for Tropical T = 293.15K and = 0.55

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TABLE 12 Ship Body_Temperature for Tropi

SUN WITH HEAVY STORM CLOUDS	VERTICAL FACE	
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330011.50011.550011.550011.550011.550011.550011.550011.550011.550011.550011.550011.950011.950011.950011.950011.950011.950011.950011.950011.950011.950011.950011.950011.950011.950011.950011.950011.950011.950011.950011.950011.950011.950011.950011.950011.950011.950011.950011.950011.950011.950011.950011.950011.950011.950011.950011.950011.950011.950011.950011.950011.950011.950011.950011.950011.950011.950011.950011.950011.950011.950011.950011.950011.950011.950011.950011.950011.950011.950011.950011.950011.950011.950011.950011.950011.950011.950011.950011.950011.950011.950011.950011.950011.950011.950011.950011.950011.950011.950011.950011.950011.950011.950011.950011.950011.950011.950011.950011.950011.950011.950011.950011.950011.950011.950011.950011.950011.950011.950011.950011.950011.950011.950011.950011.950011.950011.950011.950011.950011.950011.950011.950011.950011.950011.950011.950011.950011.950011.950011.950011.950011.950011.950011.950011.950011.950011.950011.950011.950011.950011.950011.950011.950011.950011.950011.950011.950011.950011.950011.950011.950011.950011.950011.950011.950011.950011.950011.950011.950011.950011.950011.950011.950011.950011.950011.950011.950011.950011.950011.950011.950011.950011.950011.950011.950011.950011.950011.950011.950011.950011.950011.950011.950011.950011.950011.950011.950011.950011.950011.950011.950011.950011.950011.950011.950011.950011.950011.950011.950011.950011.950011.950011.950011.950011.950011.950011.950011.950011.950011.950011.950011.950011.950011.950011.950011.950011.950011.950011.950011.950011.950011.950011.950011.950011.950011.950011.950011.950011.950011.950011.950011.950011.950011.950011.950011.950011.950011.950011.950011.950011.950011.950011.950011.950011.950011.950011.950011.950011.950011.950011.950011.950011.950011.950011.950011.950011.950011.950011.950011.950011.950011.950011.950011.950011.950011.950011.950011.950011.950011.950011.950011.950011.950011.950011.950011.950011.950011.950011.950011.950011.950011.950011.950011.950011.950011.950011.950011.950011.950011.950
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     720NTAL
CECTON TAL
120NTAL
120
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$\frac{\texttt{APPENDIX}}{\texttt{TABLES}} \ \ \underline{\texttt{B}}$ TABLES FOR DETERMINATION OF I.F.V.

13	
TABLE	

		PHI=5.00	100999888877766655544443 1009999888877766655544443 1009999888877766655544443
•	Meters	PHI=4.50	744 11100 1110
	f View in 50 m	PHI=4.00	11111111111111111111111111111111111111
	us Field or t Altitude	PHI=3.50	11111111111111111111111111111111111111
	stantaneo for Fligh	PHI=3.00	66 11111111111111111111111111111111111
	Side of In	PHI=2.50	10000000000000000000000000000000000000
		PHI=2.00	22222222222222222222222222222222222222
		DELTA	00000000000000000000000000000000000000

TABLE 14

Path in Meters for Flight Altitude= 50 m

PHI = 5 00 0 0 0 0 0 0 0 0 0 0 0 0 0 0 0 0
PHI = 4 CONTRACTOR CONTRAC
PHI = 4.00 6782 6782 6782 6786 6786 6787 6687 6687 6787 6887 6887 6887 6887 6887 6887 6887 6887 6887 6887 6887 6887 6887 6887 6887 6887
PHI = 3 .50 7774 7756777 75667777 756777 756777 75677 77567 77567 77567 7757
PHI = 3 .00
PHI = 2 10061 1004537 1004537 100120
PH = 2 12302 127902 127902 127902 127902 127903
D C C C C C C C C C C C C C C C C C C C

TABLE 15

		PHI = 5.00	11111111111111111111111111111111111111
Side of Instantaneous Field of View in Meters for Flight Altitude= 60 m	ter	PHI=4.50	55.11 100.2000 100.300 1122.201 1111 11122.001 1111 1111 111
	View in 60 m	PHI=4.00	11111111111111111111111111111111111111
	stantaneous Field for Flight Altitud	PHI=3.50	6
		PHI=3.00	2000 2000 2000 2000 2000 2000 2000 200
	ide of I	PHI=2.50	221100987764172209111009877667778809111000987768091788091111111111111111111111111111111
		PHI=2.00	08862271222222222222222222222222222222222
		DELTA	00000000000000000000000000000000000000

TABLE

PHI 95743588947743 974358874749 97435874743

 \$\frac{1}{2}\$
 \$\frac{1}{2}\$

 \$\frac{1}{2}\$ 7= 09 00 Altitude= 45556667778888690001159 45556667778888690007159 =4 888888777777777777777 12484788778877877 124877887788777 1248778877877 50 Flight 11 PHI σ for 00 Meters 11 PHI 3682546348967881479914 368273087780658147790914 Path н いめてくらわれくて多くくのわてのてわり86 11

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123617556840880496679 12361756840880496679 8027人ろびら第15の2の15のからから

		PHI=5.00	11111111111111111111111111111111111111
TABLE 17	Meters	PHI=4.50	60010000000000000000000000000000000000
	f View in 70 m	PHI=4.00	275172711111111111111111111111111111111
	us Field o t Altitude	PHI=3.50	202333227008887 20333227008889 2033327700889999 2033377700889999
	nstantaneo for Fligh	PHI=3.00	22222222222222222222222222222222222222
	Side of I	PHI=2.50	2000 1 1 1 1 1 1 1 1 1 1 1 1 1 1 1 1 1 1
•		PHI=2.00	2883/01095046651/010883 376790112833322100883 3767901283332217010883
		DELTA	00000000000000000000000000000000000000

TABLE 18

Path in Meters for Flight Altitude=70 m

PHI = 5 7777 766907 7755037 77257
PHI = 4
PHI = 4.00 955177 9951774 9929777577 992977776 992977776 9929777776 99297777777777
PHI = 3 10084 1007884 1007884 1007886 1007886 10050177 10050173 10023 10023 10023 10023 10013 10
PH I = 3 .00
PH I = 2 .5 0
PH I = 2 0 0 1 1 1 1 1 1 1 1 1 1 1 1 1 1 1 1 1
D C C C C C C C C C C C C C C C C C C C

TABLE 19

		PHI = 5.	7665544422211009982766
IADLE 17	Meters	PHI=4.50	25 25 27 27 27 27 27 27 27 27 27 27
	f View in = 80 m	PHI=4.00	22111111111111109987 000987765578078780 0009877655780780 8887806780878678
	us Field o t Altitude	PHI=3.50	22221009877669381446777757777777777777777777777777777777
	nstantaneou for Flight	PHI=3.00	8482578888630503577640 69146327109888765432100 72222222222222222222222222222222222
	Side of In	PHI=2.50	84494688867222222222222222222222222222222222
		PHI=2.00	38H222H8846775H1FFFF 398000000000000000000000000000000000000
		DELTA	00000000000000000000000000000000000000

TABLE 20

Path in Meters for Flight Altitude= 80 m

PHI=5.00	\$
PHI=4.50	88388770000 833877000000000000000000000000000000000
PHI=4.00	10092 100827.28 100827.23 10076.992 100617.727 100527.27 100109383.00 10010939393.00 1001093939393939393939393939393939393939
PHI=3.50	1122329 122329 1222329 1222329 1222329 1222329 1223232 122323 122323 122323 122323 122323 122323 122323 122323 122323 122323 122323 122323 122323 12232 1223
PHI=3.00	1239005 1239005 1239005 1239005 1239005 12390005 12390005 1239005 1239005 1239005 1239
PHI=2.50	10000000000000000000000000000000000000
PHI=2.00	22083 200663 200663 200663 2007665 20070 2
DELTA	00000000000000000000000000000000000000

21
BLE ?
TAB]

		0	24678999875WH8406H60W
		PHI=5.0(988000102020400008800 98800000000000000000000000000
	Meters	PHI=4.50	22090071 2000071 2000070 20000070 2000070 2000070 2000070 2000070 2000070 2000070 2000070 20000070 20000070 2000000 2000000 2000000 20000000 2000000
	f View in l = 90 m	PHI=4.00	2221009877998812770298877477000988774770009887747770009887747777777777
	us Field o t Altitude	PHI=3.50	2222222222224665544655466554665546603522222222222222222222222222222222222
	nstantaneou for Flight	PHI=3.00	28347111111111111111111111111111111111111
	Side of I	PHI=2.50	33333333333333333333333333333333333333
		PHI=2.00	58988511098764320976 11098764320976 11098764320976 11098764320036 11098775136
		DELTA	00000000000000000000000000000000000000

Path in Meters for Flight Altitude=90 m

PHI=5.00	298990000000000000000000000000000000000
PHI=4.50	10020000000000000000000000000000000000
PHI=4.00	12228 12228 12228 122228 122228 1221172005 11221733 11231673 11231673 11231673 11231673 11231673 11231673 11231673 11231673 11231673 1227 1227 1227 1227 1227 1227 1227 12
PHI=3.50	111229909633464772111332567389464772111222863001333464772111222863001333464772111222890096336647728873464772887347288734728874288742887428874288
PHI=3.00	100 1 1 1 1 1 1 1 1 1 1 1 1 1 1 1 1 1 1
PHI=2.50	10000000000000000000000000000000000000
PHI=2.00	30138854868883476445857 80000000000000000000000000000000000
DELTA	000000000000000000000000000000000000000

		PHI=5.00	22202440 10.1.582144322109987 10.1.58214443255470 10.1.58214443256 10.1.58214443256 10.1.5821444326 10.1.5821444326 10.1.5821444326 10.1.5821444326 10.1.5821444326 10.1.5821444326 10.1.5821444326 10.1.5821444326 10.1.5821444326 10.1.5821444326 10.1.5821444326 10.1.5821444326 10.1.5821444326 10.1.5821444326 10.1.5821444326 10.1.582144432 10.1.582144432 10.1.582144432 10.1.582144432 10.1.582144432 10.1.582144432 10.1.582144432 10.1.582144432 10.1.582144432 10.1.582144432 10.1.582144432 10.1.582144432 10.1.582144432 10.1.582144432 10.1.582144432 10.1.582144432 10.1.582144432 10.1.582144432 10.1.58214443 10.1.58214444 10.1.5821444 10.1.5821444 10.1.5821444 10.1.582144 10.1.582144 10.1.58214 10
	Meters	PHI=4.50	2222100987764443210098 222210098776443210098 322210098776443210098
	f View in =100 m	PHI=4.00	11291111111111111111111111111111111111
TABLE 23	us Field o t Altitude	PHI=3.50	2222543210 2222543210 222252222222222222222222222222222222
T	nstantaneo for Fligh	PHI=3.00	32283600074047998871 3221098765543210987571 33210998765570570586271
	Side of I	PHI=2.50	38245539946532908704578282 3005432965329108767 3005432003060850465282
		PHI = 2.00	17.6081027642198 19.752074073950504643198 19.752074073950504643198 19.75207407394643198
		DELTA	00000000000000000000000000000000000000

	PHI=5.00	10090000000000000000000000000000000000
m 0	PHI=4.50	10000000000000000000000000000000000000
Altitude=100	PHI=4.00	023558849800044987951113555778849890011111111111111111111111111111111
r Flight A	PHI=3.50	11111111111111111111111111111111111111
Meters for	PHI=3.00	17.29 17.29
Path in	PHI=2.50	11111111111111111111111111111111111111
	PHI=2.00	22222222222222222222222222222222222222
	DELTA	00000000000000000000000000000000000000

$\frac{\texttt{APPENDIX}}{\texttt{TABLES}} \; \frac{\texttt{C}}{\texttt{VITH}} \; \; \texttt{DATA} \; \; \texttt{USED} \; \; \texttt{FOR} \; \; \texttt{LOWTRAN} \; \; \texttt{6}$

TABLE 25 Data Used for Month October

MODEL ITYPE IEMSCT M1 M2 M3 IM NOPRT TBOUND SALB	2 : Midlatitude Summer T and P Profile 2 : Midlatitude Summer Water Vapor Profile 2 : Midlatitude Summer Ozone Profile 0 : Normal Operation of Program 0 : Normal Operation of Program
	CARD 2 : 0 : Normal Operation of Program CARD 2 : 3 : Navy Maritime Extinction : 2 : Fall - Winter : 0 : No Stratospheric Background : 5 : Medium Continental Influence : 0 : No Cirrus : 0 : Not Used : 10 : Meteorological Range(Km) : 2.09: Current Wind Speed (m/s) : 2.09: 24 hours Average Wind Speed (m/s) : 0 : Rain Rate (mm/h) CARD 3
H1 H2 ANGLE RANGE BETA RO LEN	: From Section G of Chapter II : Initial Altitude : From Table 22 : Final Altitude : : Not Necessary to Define : From Table 22 : Path Length : : Not Necessary to Define : : Not Necessary to Define : : Not Necessary to Define : 0 : Normal Operation CARD 4
V1 V2 DV	: 2255 : Initial Frequency : 2000 : Final Frequency : 10 : Frequency Increment

	Average	T ge Transmitt	Transmittances for Month October	nth October		
FINAL ALTITUDE	0.050	0.060	0.070	0.080	0.090	0.100
INITIAL	(Km)	(Km)	(Кш)	(Km)	(Km)	(Km)
0.004 (Km)	0.4718	0.4715	0.4712	0.4712	0.4711	0.4711
0.007 (km)	0.4719	0.4716	0.4713	0.4713	0.4712	0.4712
0.012 (Km)	0.4720	0.4717	0.4715	0.4715	0.4714	0.4714
0.016 (Km)	0.4722	0.4719	0.4716	0.4716	0.4715	0.4715
0.018 (Km)	0.4722	0.4719	0.4717	0.4717	0.4716	0.4716
0.021 (Km)	0.4723	0.4720	0.4718	.0.4718	0.4717	0.4717
0.022 (Km)	0.4723	0.4720	0.4718	0.4718	0.4717	0.4717
Average per Altitude	0.4721	0.4718	0.4716	0.4716	0.4715	0.4715
	Tot	Total Average of	of the Month	1 = 0.4717		

TABLE 27 Data Used for Month November

MODEL ITYPE IEMSCT M1 M2 M3 IM NOPRT TBOUND SALB	2 : Midlatitude Summer 2 : Slant Path Between Two Altitudes 0 : Program Execution in Transmittance Mode 2 : Midlatitude Summer T and P Profile 2 : Midlatitude Summer Water Vapor Profile 2 : Midlatitude Summer Ozone Profile 0 : Normal Operation of Program CARD 2
IHAZE ISEASN IVULCN ICSTL ICIR IVSA VIS WSS WHH RAINRT	3 : Navy Maritime Extinction 2 : Fall - Winter 0 : No Stratospheric Background 5 : Medium Continental Influence 0 : No Cirrus 0 : Not Used 10 : Meteorological Range (Km) 2.09: Current Wind Speed(m/s) 2.09: 24 hours Average Wind Speed (m/s) 0 : Rain Rate (mm/h) CARD 3
H1 H2 ANGLE RANGE BETA RO LEN	From Section G of Chapetr II: Initial Altitude Form Table 22: Final Altitude: Not Necessary to Define From Table 22: Path Length: Not Necessary to Define: Not Necessary to Define: Not Necessary to Define: Normal Operation CARD 4
V1 V2 DV	: 2255 : Initial Frequency : 2000 : Final Frequency : 10 : Frequency Increment

	0 0 0 T 0 T 0 T 0 T 0 T 0 T 0 T 0 T 0 T	+ t mo re c + T	28 for	Month November		
FINAL	0.050		070	0.080	. 060.0.	0.100
INITIAL. ALTITUDE	(Km)	(Km)	(Km)	(Km)	(Km)	(Km)
0.004 (Km)	0.4705	0.4702	0.4699	6694.0	0.4698	0.4698
0.007 (Km)	0.4706	0.4703	0.4700	0.4700	0.4699	0.4699
0.012 (Km)	0.4708	0.4705	0.4702	0.4702	0.4701	0.4701
0.016 (Km)	0.4709	0.4706	0.4703	0.4703	0.4702	0.4702
0.018 (Km)	0.4710	0.4707	0.4704	40.4.0	0.4703	0.4703
0.021 (Km)	0.4711	0.4708	0.4705	0.4705	0.4704	0.4704
0.022 (Km)	0.4711	8074.0	0.4705	0.4705	0.4704	0.4704
Average per Altitude	0.4709	0.4706	0.4703	0.4703	0.4702	0.4702
	Total	Average	of the Month	- 0.4704		1

Data Used for Month December

MODEL ITYPE IEMSCT M1 M2 M3 IM NOPRT TBOUND SALB	: 0 : Program Execution in Transmittance Mode : 2 : Midlatitude Summer T and P Profile : 2 : Midlatitude Summer Water Vapor Profile : 2 : Midlatitude Summer Ozone Profile : 0 : Normal Operation of Program : 0 : Normal Operation of Program
IHAZE ISEASN	3 : Navy Maritime Extinction 2 : Fall - Winter 0 : No Stratospheric Background 5 : Medium Continental Influence 0 : No Cirrus 0 : Not Used 10 : Meteorological Range 2.05: Current Wind Speed 2.05: Current Wind Speed 2.05: 24 hours Average Wind Speed (m/s)
IVULCN ICSTL ICIR	: 0 : No Stratospheric Background : 5 : Medium Continental Influence : 0 : No Cirrus
IVSA VIS	: 0 : Not Used : 10 : Meteorological Range : 2 05: Current Wind Speed
WHH RAINRT	: 2.05: 24 hours Average Wind Speed (m/s) : 0 : Rain Rate (mm/h)
	CARD 3
H1 H2 ANGLE RANGE BETA RO LEN	: From Section G of Chapter II : Initial Altitude : From Table 22 : Final Altitude : : Not Necessary to Define : From Table 22 : Path Length : : Not Necessary to Define : : Not Necessary to Define : 0 : Normal Operation
2211	CARD 4
V1 V2 DV	: 2255 : Initial Frequency : 2000 : Final Frequency : 10 : Frequency Increment

TANT	Average	1	TABLE 30 Transmittances for Month December	onth December		
ALTITUDE	0.050	0.060	0.070	0.080	060.0	0.100
INITIAL ALTITUDE	(Km)	(Km)	(Кл.)	(Km)	(Km)	(Km)
0.004 (Km)	0.4705	0.4702	6694.0	0.4699	0.4698	0.4698
0.007 (Km)	0.4706	0.4703	0.4700	0.4700	0.4699	0.4699
0.012 (Km)	0.4708	0.4705	0.4702	0.4702	0.4701	0.4701
0.016 (Km)	0.4709	0.4706	0.4703	0.4703	0.4702	0.4702
0.018 (Km)	0.4710	0.4707	0.4704	0.4704	0.4703	0.4703
0.021 (Km)	0.4711	0.4708	0.4705	0.4705	0.4704	0.4704
0.022 (Km)	0.4711	0.4708	0.4705	0.4705	0.4704	0.4704
Average per Altitude	0.4709	0.4706	0.4703	0.4703	0.4702	0.4702
	Tot	Total Average	of the Month	+024.0 =		

Data Used for Month January

MODEL ITYPE IEMSCT M1 M2 M3 IM NOPRT TBOUND SALB	2 : Midlatitude Summer 2 : Slant Path Between Two Altitudes 0 : Program Execution in Transmittance Mode 2 : Midlatitude Summer T and P Profile 2 : Midlatitude Summer Water Vapor Profile 2 : Midlatitude Summer Ozone Profile 3 : Midlatitude Summer Ozone Profile 4 : Normal Operation of Program 5 : Normal Operation of Program 6 : Normal Operation of Program 7 : O : Normal Operation of Program 8 : O : Normal Operation of Program 9 : Normal Operation of Program 10 : Normal Operation of Program 11 : O : Normal Operation of Program 12 : O : Normal Operation of Program
IHAZE ISEASN IVULCN ICSTL ICIR IVSA VIS WSS WHH RAINRT	3 : Navy Maritime Extinction 2 : Fall - Winter 0 : No Stratospheric Background 5 : Medium Continental Influence 0 : No Cirrus 0 : Not Used 15 : Meteorological Range 2.06: Current Wind Speed 2.06: 24 hours Average Wind Speed (m/s) 0 : Rain Rate (mm/h)
H1 H2 ANGLE RANGE BETA RO LEN	: From Section G of Chapter II : Initial Altitude : From Table 22 : Final Altitude : : Not Necessary to Define : From Table 22 : Path Length : : Not Necessary to Define : : Not Necessary to Define : : Not Necessary to Define : 0 : Normal Operation CARD 4
V1 V2 DV	: 2255 : Initial Frequency : 2000 : Final Frequency : 10 : Frequency Increment

	-	Averag	T e Transmitt	TABLE 32 Average Transmittances for Month January	nth January,		
FINAL ALTITUDE)E 0.050	0	0.060	0.070	0.080	. 060.0	0.100
INITIAL. ALTITUDE	(Km)		(Km)	(Km)	(Km)	(Km)	(Km)
0.004 (Km)	0.4705	7 0 5	0.4702	0.4699	0.4699	0.4698	0.4698
0.007 (Km)	0.4706	902	0.4703	0.4700	0.4700	0.4699	0.4699
0.012 (Km)	0.4708	807	0.4705	0.4702	0.4702	0.4701	0.4701
0.016 (Km)	0.4709	709	0.4706	0.4703	0.4703	0.4702	0.4702
0.018 (Km)	0.4710	710	0.4707	0.4704	0.4704	0.4703	0.4703
0.021 (Km)	0.4711	711	0.4708	0.4705	0.4705	0.4704	0.4704
0.022 (Km)	0.471	111	0.4708	0.4705	0.4705	0.4704	0.4704
Average per Altitude	0.4709	607	0.4706	0.4703	0.4703	0.4702	0.4702
		Total	Average	of the Month	4064.0 =		

Data Used for Month February

MODEL ITYPE IEMSCT M1 M2 M3 IM NOPRT TBOUND SALB	2 : Midlatiitude Summer 2 : Slant Path Between Two Altitudes 0 : Program Execution in Transmittance Mode 2 : Midlatitude Summer T and P Profile 2 : Midlatitude Summer Water Vapor Profile 2 : Midlatitude Summer Ozone Profile 3 : Midlatitude Summer Ozone Profile 4 : Mormal Operation of Program 5 : Normal Operation of Program 6 : Normal Operation of Program 7 : Normal Operation of Program 8 : O : Normal Operation of Program 9 : Normal Operation of Program 9 : Normal Operation of Program 9 : Normal Operation of Program
IHAZE ISEASN IVULCN ICSTL ICIR IVSA VIS WSS WHH RAINRT	: 3 : Navy Maritime Extinction : 2 : Fall - Winter : 0 : No Stratospheric Background : 5 : Medium Continental Influence : 0 : No Cirrus : 0 : Not Used : 15 : Meteorological Range (Km) : 2.22: Current Wind Speed (m/s) : 2.22: 24 hours Average Wind Speed (m/s) : 0 : Rain Rate (mm/h) CARD 3
H1 H2 ANGLE RANGE BETA RO LEN	: From Section G of Chapter II : Initial Altitude : From Table 22 : Final Altitude : : Not Necessary to Define : From Table 22 : Path Length : : Not Necessary to Define : : Not Necessary to Define : : Not Necessary to Define : 0 : Normal Operation CARD 4
V1 V2 DV	: 2255 : Initial Frequency : 2000 : Final Frequency : 10 : Frequency Increment

	Average	TA e Transmitta	TABLE 34 Transmittances for Month February	nth February		
FINAL	0:00	0.060	0.070	0.080	060.0	0.100
INITIAL ALTITUDE	(Km)	(Km)	(Km)	(Km)	(Km)	(Km)
0.004 (Km)	0.4711	0.4708	0.4705	4074.0	0.4704	0.4704
0.007 (Km)	0.4712	0.4709	0.4706	0.4705	0.4705	0.4705
6.012 (Km)	0.4714	0.4711	0.4708	0.4707	0.4707	0.4707
0.016 (Km)	0.4715	0.4712	0.4709	0.4708	0.4708	0.4708
0.018 (Km)	0.4716	0.4713	0.4710	0.4709	0.4709	0.4709
0.021 (Km)	0.4717	4174.0	0.4712	0.4710	0.4710	0.4710
0.022 (Km)	0.4717	0.4714	0.4711	0.4710	0.4710	0.4710
Average per Altitude	0.4715	0.4712	0.4709	0.4708	0.4708	0.4708
	Tot	Total Average o	of the Month	= 0.4710		

Data Used for Month March

MODEL ITYPE IEMSCT M1 M2 M3 IM NOPRT TBOUND SALB	: 2 : Midlatitude Summer : 2 : Slant Path Between Two Altitudes : 0 : Program Execution in Transmittance Mode : 2 : Midlatitude Summer T and P Profile : 2 : Midlatitude Summer Water Vapor Profile : 2 : Midlatitude Summer Ozone Profile : 0 : Normal Operation of Program : CARD 2
IHAZE ISEASN IVULCN ICSTL ICST IVSA VIS WSS WHH RAINRT	3 : Navy Maritime Extinction 2 : Fall - Winter 0 : No Stratospheric Background 5 : Medium Continental Influence 0 : No Cirrus 0 : Not Used 15 : Meteorological Range (Km) 2.13: Current Wind Speed (m/s) 2.13: 24 hours Average Wind Speed (m/s) 0 : Rain Rate (mm/h) CARD 3
H1 H2 ANGLE RANGE BETA RO LEN	: From Section G of Chapter II : Initial Altitude : From Table 22 : Final Altitude : : Not Necessary to Define : From Table 22 : Path Length : : Not Necessary to Define : : Not Necessary to Define : : Not Necessary to Define : 0 : Normal Operation CARD 4
V1 V2 DV	: 2255 : Initial Frequency : 2000 : Final Frequency : 10 : Frequency Increment

	90 0.100	711. 0.4711	712 0.4712	114 0.4714	0.4715	0.4716	717 0.4717	717 0.4717	715 0.4715	
	0.090 (Km)	0.4711	0.4712	0.4714	0.4715	0.4716	0.4717	0.4717	0.4715	
5 for Month March	0.080 (Km)	0.4711	0.4712	0.4714	0.4715	0.4716	0.4717	0.4717	0.4715	= 0.4713
	0.070 (Km)	0.4712	0.4713	0.4715	0.4716	0.4717	0.4718	0.4718	0.4716	of the Month =
Transmi	0.060	0.4713	0.4714	0.4716	0.4717	0.4718	0.4719	0.4719	0.4717	Average
Average	0.050 (Km)	0.4718	0.4719	0.4720	0.4722	0.4722	0.4723	0.4723	0.4721	Total
	ALTITUDE ALTITUDE	rube (Km)	(Km)	(Km)	(Km)	(Km)	(Km)	(Km)	e per de	
	FINAL	ALTITUDE	0.007	0.012	0.016	0.018	0.021	0.022	Average Altitude	

Data Used for Month April

MODEL ITYPE IEMSCT M1 M2 M3 IM NOPRT TBOUND SALB	2 : Midlatitude Summer 2 : Slant Path Between Two Altitudes 0 : Program Execution in Transmittance Mode 2 : Midlatitude Summer T and P Profile 2 : Midlatitude Summer Water Vapor Profile 2 : Midlatitude Summer Ozone Profile 3 : Midlatitude Summer Ozone Profile 4 : O : Normal Operation of Program 5 : Normal Operation of Program 6 : Normal Operation of Program 7 : O : Normal Operation of Program 8 : O : Normal Operation of Program 9 : O : Normal Operation of Program 10 : Normal Operation of Program 11 : O : Normal Operation of Program 12 : O : Normal Operation of Program
IHAZE ISEASN IVULCN ICSTL ICIR IVSA VIS WSS WHH RAINRT	: 3 : Navy Maritime Extinction : 2 : Fall - Winter : 0 : No Stratospheric Background : 5 : Medium Continental Influence : 0 : No Cirrus : 0 : Not Used : 15 : Meteorological Range (Km) : 2.12 : Current Wind Speed (m/s) : 2.12 : 24 hours Average Wind Speed (m/s) : 0 : Rain Rate (mm/h) CARD 3
H1 H2 ANGLE RANGE BETA RO LEN	: From Section G of Chapter II : Initial Altitude : From Table 22 : Final Altitude : : Not Necessary to Define : From Table 22 : Path Length : : Not Necessary to Define : : Not Necessary to Define : : Not Necessary to Define : 0 : Normal Operation CARD 4
V1 V2 DV	: 2255 : Initial Frequency : 2000 : Final Frequency : 10 : Frequency Increment

TABLE 38 Average Transmittances for Month April	ALTITUDE 0.050 0.060 0.070 0.080 0.090 0.100 (Km) (Km) (Km) (Km)	4 (Km) 0.4718 0.4713 0.4712 0.4711 0.4711 0.4711	7 (Km) 0.4719 0.4714 0.4713 0.4712 0.4712 0.4712	2 (Km) 0.4720 0.4716 0.4715 0.4714 0.4714 0.4714	3 (Km) 0.4722 0.4717 0.4716 0.4715 0.4715 0.4715	3 (Km) 0.4722 0.4718 0.4717 0.4716 0.4716 0.4716	1 (Km) 0.4723 0.4719 0.4718 0.4717 0.4717 0.4717	2 (Km) 0.4723 0.4719 0.4718 0.4717 0.4717 0.4717	ge per 0.4721 0.4717 0.4716 0.4715 0.4715 0.4715	Total Avenage of the Month = 0.4713
	FINAL ALTITUDE INITIAL ALTITUDE	0.004 (Km)	0.007 (Km)	0.012 (Km)	0.016 (Km)	0.018 (Km)	0.021 (Km)	0.022 (Km)	Average per Altitude	

TABLE 39 Data Used for Month May

MODEL ITYPE IEMSCT M1 M2 M3 IM NOPRT TBOUND SALB	: 1 : Tropical Atmosphere : 2 : Slant Path Between Two Altitudes : 0 : Program Execution in Transmittance Mode : 1 : Tropical Temperature and Presure Profile : 1 : Tropical Water Vapor Profile : 1 : Tropical Ozone Profile : 0 : Normal Operation of Program
IHAZE ISEASN IVULCN ICSTL ICIR IVSA VIS WSS WHH RAINRT	: 3 : Navy Maritime Extinction : 1 : Spring - Summer : 0 : No Stratospheric Background : 5 : Medium Continental Influence : 0 : No Cirrus : 0 : Not Used : 15 : Meteorological Range (Km) : 2.02: Current Wind Speed (m/s) : 2.02: 24 hours Average Wind Speed (m/s) : 0 : Rain Rate (mm/h) CARD 3
H1 H2 ANGLE RANGE BETA RO LEN	: From Section G of Chapter II : Initial Altitude : From Table 22 : Final Altitude : : Not Necessary to Define : From Table 22 : Path Length : : Not Necessary to Define : : Not Necessary to Define : : Not Necessary to Define : 0 : Normal Operation CARD 4
V1 V2 DV	: 2255 : Initial Frequency : 2000 : Final Frequency : 10 : Frequency Increment

	Ave	Average Transmi	TABLE 40 Transmittances for	for Month May		
FINAL	0:000	090.0	0.070	0.080	0.090	0.100
INITIAL ALTITUDE	(Km)	(Km)	(Km)	(Km)	(Km)	(Km)
0.004 (Km)	0.4270	0.4267	0.4264	0.4262	0.4263	0.4263
0.007 (Km)	0.4271	0.4268	0.4265	0.4263	0.4264	0.4264
0.012 (Km)	0.4273	0.4270	0.4267	0.4265	0.4266	0.4266
0.016 (Km)	0.4274	0.4271	0.4268	0.4267	0.4268	0.4268
0.018 (Km)	0.4275	0.4272	0.4269	0.4268	0.4269	. 0.4269
0.021 (Km)	0.4276	0.4273	0.4270	0.4269	0.4270	0.4270
0.022 (Km)	0.4276	0.4273	0.4270	0.4269	0.4270	0.4270
Average per Altitude	0.4273	0.4271	0.4268	0.4266	0.4267	0.4267
,	Total	Average	of the Month	= 0.4269		

Data Used for Month June

CARD 1

IM : 0 : Normal Operation of Program NOPRT : 0 : Normal Operation of Program TBOUND : 0 : Normal Operation of Program SALB : 0 : Normal Operation of Program	TBOUND		0 1 1 0 0		Normal Operation of Program
--	--------	--	-----------	--	-----------------------------

CARD 2

CARD 3

H1	: From Section G of Chapter II : Initial Altitude
H2	: From Table 22 : Final Altitude
ANGLE	: : Not Necessary to Define
RANGE	: From Table 22 : Path Length
BETA	: : Not Necessary to Define
RO	: : Not Necessary to Define
LEN	: 0 : Normal Operation
	· · · · · · · · · · · · · · · · · · ·

<u>V2</u> : 2000 : I	Initial Frequency Final Frequency Frequency Increment
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		Ę-	TART 4.2			
	Ave	Average Transmittances		for Month June		
FINAL	0.050	090.0	0.070	0.080	0.090	0.100
INITIAL ALTITUDE	(Km)	(Km)	(Km)	(Кш)	(Km)	(Km)
0.004 (Km)	0.4264	0.4261	0.4258	0.4257	0.4257	0.4257
0.607 (Km)	0.4265	0.4262	0.4259	0.4258	0.4258	0.4258
0.012 (Km)	0.4267	0.4264	0.4261	0.4260	0.4260	0.4260
0.016 (Km)	0.4268	0.4265	0.4262	0.4262	0.4262	0.4262
0.018 (Km)	0.4269	0.4266	0.4263	0.4262	0.4262	0.4262
0.021 (Km)	0.4270	0.4267	0.4264	0.4263	0.4263	0.4263
0.022 (Km)	0.4270	0.4267	0.4264	0.4263	0.4263	0.4263
Average per Altitude	0.4267	0.4264	0.4261	0.4260	0.4260	0.4260
	Tot	al Average o	Total Average of the Month = 0.4262	= 0.4262		

TABLE 43 Data Used for Month July

MODEL ITYPE IEMSCT M1 M2 M3 IM NOPRT TBOUND SALB	: 1 : Tropical Atmosphere : 2 : Slant Path Between Two Altitudes : 0 : Program Execution in Transmittance Mode : 1 : Tropical Temperature and Presure Profile : 1 : Tropical Water Vapor Profile : 1 : Tropical Ozone Profile : 0 : Normal Operation of Program
IHAZE ISEASN IVULCN ICSTL ICIR IVSA VIS WSS WHH RAINRT	: 3 : Navy Maritime Extinction : 1 : Spring - Summer : 0 : No Stratospheric Background : 5 : Medium Continental Influence : 0 : No Cirrus : 0 : Not Used : 12 : Meteorological Range (Km) : 2.15: Current Wind Speed (m/s) : 2.15: 24 hours Average Wind Speed (m/s) : 0 : Rain Rate (mm/h) CARD 3
H1 H2 ANGLE RANGE BETA RO LEN	: From Section G of Chapter II : Initial Altitude : From Table 22 : Final Altitude : : Not Necessary to Define : From Table 22 : Path Length : : Not Necessary to Define : : Not Necessary to Define : : Not Necessary to Define : 0 : Normal Operation CARD 4
V1 V2 DV	: 2255 : Initial Frequency : 2000 : Final Frequency : 10 : Frequency Increment

	Ave	T Average Transmi	TABLE 44 Transmittances for	for Month July		
FINAL	0.050	090.0	0.070	0.080	0.60.0	0.100
INITIAL ALTITUDE	(Km)	(Km)	(Km)	(Km)	(Km)	(Km)
0.004 (Km)	. 0.4264	0.4261	0.4258	0.4257	0.4257	0.4257
0.007 (Km)	0.4265	0.4262	0.4259	0.4258	0.4258	0.4258
0.012 (Km)	0.4267	0.4264	0.4261	0.4260	0.4260	0.4260
0.016 (Km)	0.4268	0.4265	0.4262	0.4262	0.4262	0.4262
0.018 (Km)	0.4269	0.4266	0.4263	0.4262	0.4262	0.4262
0.021 (Km)	0.4270	0.4267	0.4264	0.4263	0.4263	0.4263
0.022 (Km)	0.4270	0.4267	0.4264	0.4263	0.4263	0.4263
Average per Altitude	0.4267	0.4264	0.4261	0.4260	0.4260	0.4260
1	Total	Average	of the Month	= 0.4262		

TABLE 45

Data Used for Month August

CARD 1

MODEL ITYPE IEMSCT M1 M2 M3 IM NOPRT TBOUND SALB	l : Tropical Atmosphere 2 : Slant Path Between Two Altitudes 0 : Program Execution in Transmittance Mode 1 : Tropical Temperature and Presure Profile 1 : Tropical Water Vapor Profile 1 : Tropical Ozone Profile 0 : Normal Operation of Program CARD 2
IHAZE ISEASN IVULCN ICSTL ICIR IVSA VIS WSS WHH RAINRT	3 : Navy Maritime Extinction 1 : Spring - Summer 0 : No Stratospheric Background 5 : Medium Continental Influence 0 : No Cirrus 0 : Not Used 10 : Meteorological Range (Km) 2.20 : Current Wind Speed (m/s) 2.20 : 24 hours Average Wind Speed (m/s) 0 : Rain Rate (mm/h) CARD 3
H1 H2 ANGLE RANGE BETA RO LEN	: From Section G of Chapter II : Initial Altitude : From Table 22 : Final Altitude : : Not Necessary to Define : From Table 22 : Path Length : : Not Necessary to Define : : Not Necessary to Define : 0 : Normal Operation CARD 4
V1 V2 DV	: 2255 : Initial Frequency : 2000 : Final Frequency : 10 : Frequency Increment

	Average	Transmi		5 for Month August		
FINAL ALTITUDE INITIAL	0.050 (Km)	0.060 (Km)	0.070 (Km)	0.080 (Km)	0.090 (Km)	0.100 (Km)
ALTITUDE 0.004 (Km)	0.4258	0.4255	0.4252	0.4251	0.4251	0.4251
0.007 (Km)	0.4259	0.4256	0.4253	0.4252	0.4252	0.4252
0.012 (Km)	0.4261	0.4258	0.4255	0.4253	0.4253	0.4253
0.016 (Km)	0.4262	0.4259	0.4256	0.4255	0.4255	0.4255
0.018 (Km)	0.4263	0.4260	0.4257	0.4256	0.4256	0.4256
0.021 (Km)	0.4264	0.4261	0.4258	0.4257	0.4257	0.4257
0.022 (Km)	0.4264	0.4261	0.4258	0.4257	0.4257	0.4257
Average per Altitude	0.4261	0.4258	0.4255	0.4254	0.4254	0.4254
	TOtal	Average	of the Month	= 0.4256		

TABLE 47 Data Used for Month September

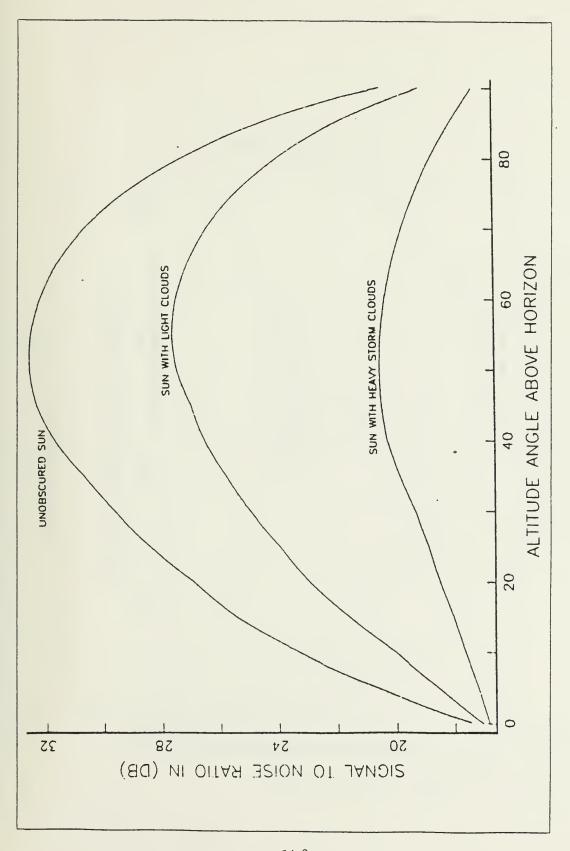
CARD 1

MODEL ITYPE IEMSCT M1 M2 M3 IM NOPRT TBOUND SALB	: l : Tropical Atmosphere : 2 : Slant Path Between Two Altitudes : 0 : Program Execution in Transmittance Mode : l : Tropical Temperature and Presure Profile : l : Tropical Water Vapor Profile : l : Tropical Ozone Profile : l : Tropical Ozone Profile : 0 : Normal Operation of Program : CARD 2
IHAZE ISEASN IVULCN ICSTL ICIR IVSA VIS WSS WHH RAINRT	<pre>3 : Navy Maritime Extinction 1 : Spring - Summer 0 : No Stratospheric Background 5 : Medium Continental Influence 0 : No Cirrus 0 : Not Used 8 : Meteorological Range (Km) 2.17: Current Wind Speed (m/s) 2.17: 24 hours Average Wind Speed (m/s) 0 : Rain Rate (mm/h)</pre>
H1 H2 ANGLE RANGE BETA RO LEN	: From Section G of Chapter II : Initial Altitude : From Table 22 : Final Altitude : : Not Necessary to Define : From Table 22 : Path Length : : Not Necessary to Define : : Not Necessary to Define : : Not Necessary to Define : 0 : Normal Operation CARD 4
V1 V2 DV	: 2255 : Initial Frequency : 2000 : Final Frequency : 10 : Frequency Increment

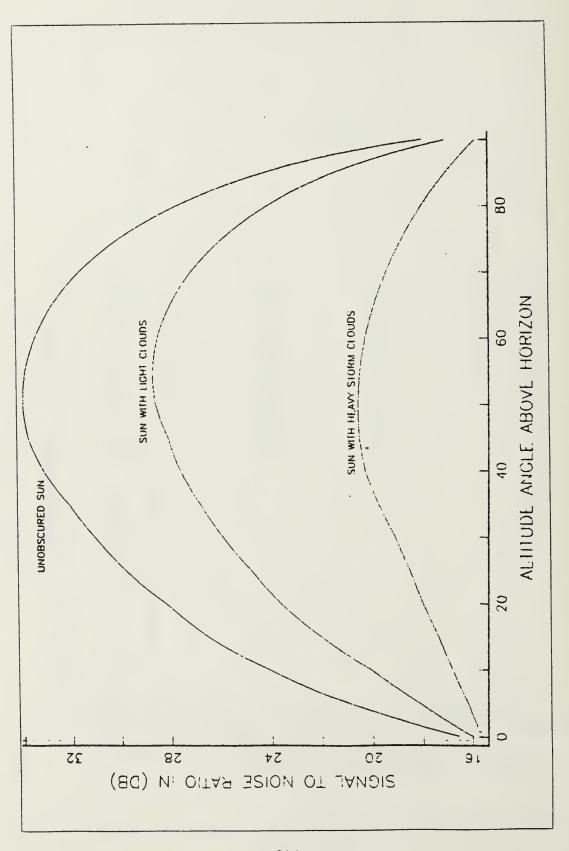
	0.100	(Km)	0.4242	0.4243	0.4245	0.4246	0.4247	0.4248	0.4248	0.4245	
	060.0	(Km)	0.4242	0.4243	0.4245	0.4246	0.4247	0.4248	0.4248	0.4245	
48 for Month September	0.080	(Km)	0.4242	0.4243	0.4245	0.4246	0.4247	0.4248	0.4248	0.4245	= 0.4247
1	0.070	(Km)	0.4243	ካቱሪቱ°0	0.4246	0.4247	0.4248	0.4249	0.4249	0.4246	Average of the Month
Transmitt	090.0	(Km)	0.4246	0.4247	0.4249	0.4250	0.4251	0.4252	0.4252	0.4249	
Average	0.050	(Km)	0.4249	0.4250	0.4252	0.4253	0.4254	0.4255	0.4255	0.4252	Total
	FINAL	INITIAL ALTITUDE	0.004 (Km)	G.007 (Km)	0.012 (Km)	0.016 (Km)	0.618 (Km)	. 0.021 (Km)	0.022 (Km)	Average per Altitude	

APPENDIX D PLOTS OF S/N RATIO

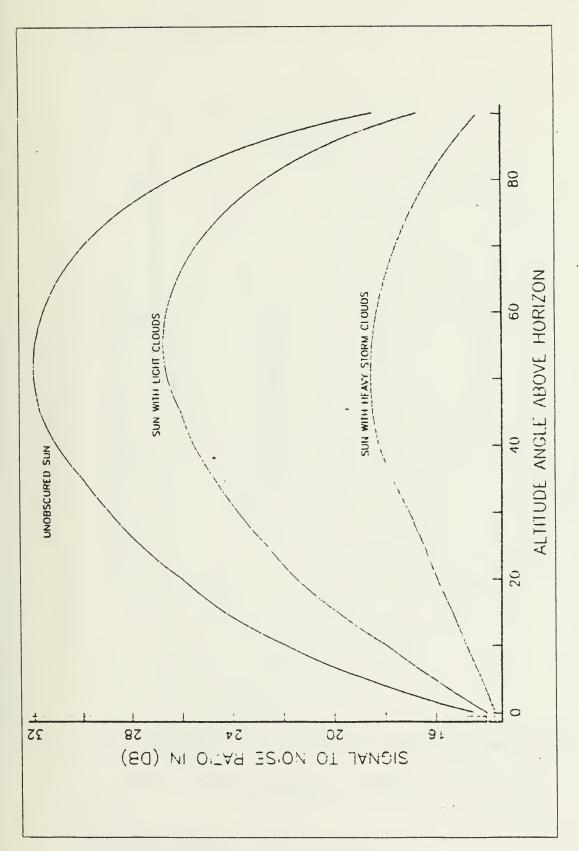
S/N Ratio for Midlatitude Summer, 6.=0.80 and Flight Altitude 50m Figure D.1



S/N Ratio for Midlatitude Summer, &=0.80 and Flight Altitude 100m Figure D.2



S/N Ratio for Midlatitude Summer, £=0.55 and Flight Altitude 50m Figure D.3



=0.55 S/N Ratio for Midlatitude Summer, and Flight Altitude 100m Figure D.4

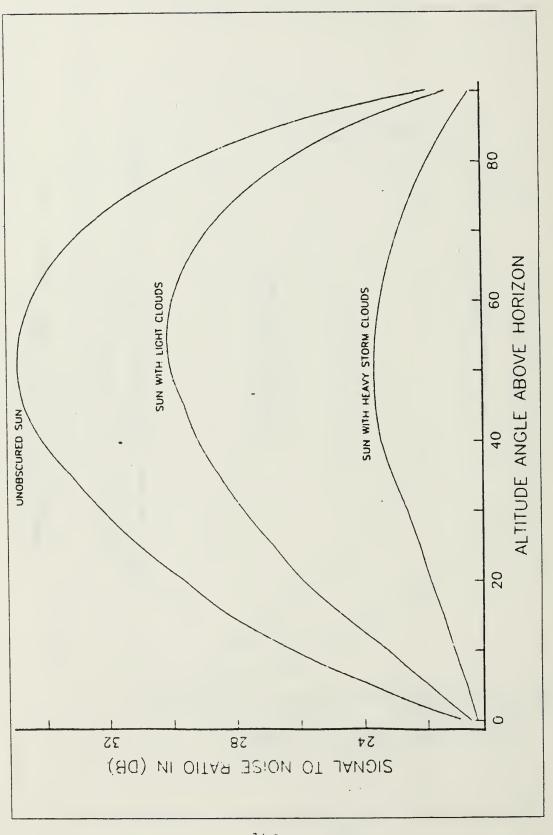


Figure D.5 S/N Ratio for Tropical, &=0.80 and Flight Altitude 50m

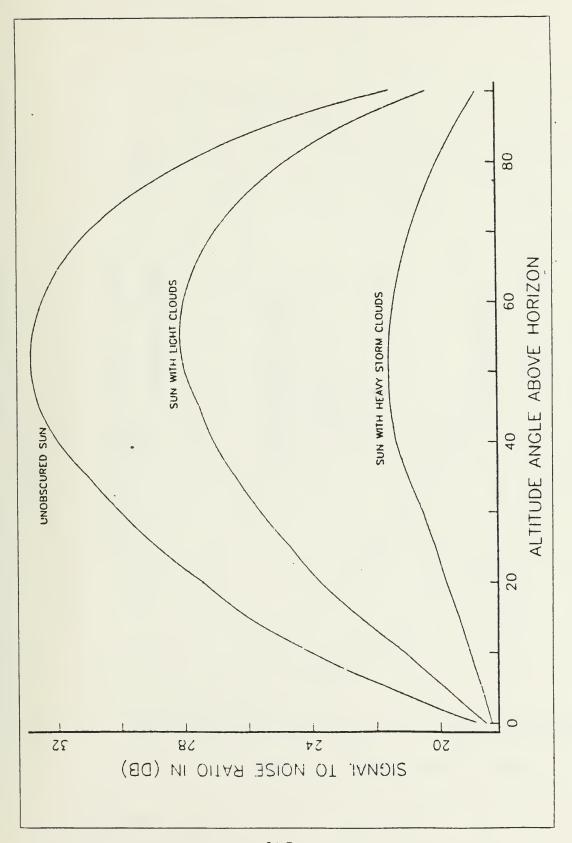


Figure D.6 S/N Ratio for Tropical, &=0.80 and Flight Altitude 100m

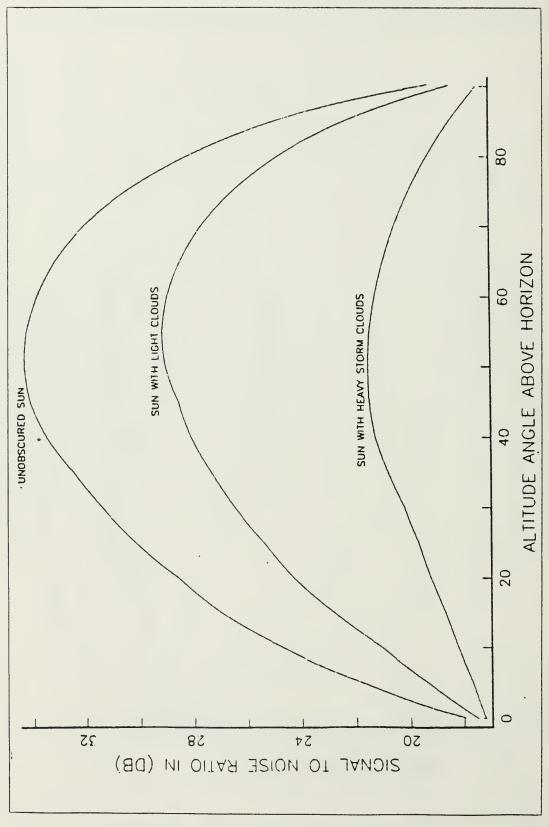


Figure D.7 S/N Ratio for Tropical, &=0.55 and Flight Altitude 50m

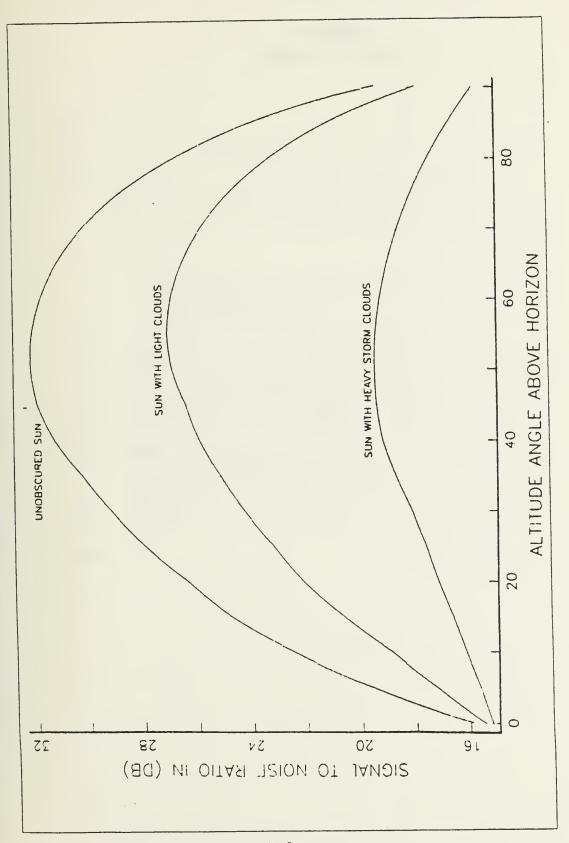


Figure D.8 S/N Ratio for Tropical, =0.5: and Flight Altitude 100m

APPENDIX E COMPUTER PROGRAMS

```
$ J O B
                                                                    PROGRAM SHIP1
               CALCULATION OF SIDE OF INSTANTANEOUS FIELD OF VIEW
                           DIMENSION H(6), PHI(7), DELTA(21), SIDE(6,7,21), PATH(6,7,21)
 Ç
                   DATA (H(I), I=1,6)/50.,60.,70.,80.,30.,100./
DATA (PHI(K),6=1,7)/2.0,2.5,3.0,3.5,4.0,4.5,5.0/
DATA (DELIA(L),L=1,21)/.2...22...24...26...23...32...34...36...38.
1 .4...42...44...46...43...5...52...54...56...58...0/
 C
                  DO 50 I=1,6

DO 45 K=1,7

DO 40 L=1,21

TH= (90.-PHI(K) -DELTA(L)) *3.141592/100.

B=d(I) *IAN(IH)

C=(H(I) *H(I) +b*H) **.5

D=DE_TA(L) *3.141592/100.

SIDE(I, K, L) = 2.*C *SIN(E)

PATH(I, K, L) = SIDE(I, K, L)/(2.*TAN(E))

CONTINUE

CONTINUE

WRITE(0, 39) H(I)

FURMAT('I'H 4K, 'SIDE OF INSTANTANEOUS FIELD OF VIEW IN METERS'//

PORMAT('I'H 4K, 'SIDE OF INSTANTANEOUS FIELD OF VIEW IN METERS'//

BURITE(0, 30) (PHI(X), K=1,7)

PORMAT(2K, 'DELTA', 3K, 7('PHI=', F4.2, 2X)/)

DO 35 L=1,21

WRITE(0, 34) DELTA(L), (SIDE(I, K, L), K=1,7)

FORMAT('O', 2X, F4.2, 1X, 7F10.2)

CONTINUE

HETTE(0, 49) H(I)

HETTE(0, 49) H(I)
         40
         39
         38
                                  WPITZ(6,49) H(I)
PORMAT(11,4% PATH IN DETERS*//5%, FOR FLIGHT HEIGHT=*,F4.J,
1% (METERS*//)
PORMAT(2,43) (PHI(K),K=1,7)
PORMAT(2,43) (PHI(K),K=1,7)
DO 47 L=1,21
TRITE(6,40) DELIA(L),(PATH(I,K,L),K=1,7)
FORMAT(10,2%,F4.2,1%,7F1).2)
CONTINUE
CONTINUE
STCP
END
         49
         48
         46
47
50
  SENTRY
```

```
$10B
C
C
                                                                                                                                                                                                                                               PROGRAM SHIP THERMAL SIGNATURE
                                                                                                                                                                                                                                                                           T(2,2,3,91), TIN(2), TS(3,91), ESN(2,3,91), D(91)

BDL(2,2,3,91), ADLR(2,3,91), HDLT(2,2,3,91)

BI1(3,91), AI2(3,91), AI3(3,91), BI4(3,91)

KI5(3,91), SI7(3,91)

VS(3,91), SNR(3,91)
                                                                                                     DIMENSION
DIMENSION
DIMENSION
DIMENSION
DIMENSION
                                                                                            DATA (TIN(J), J=1,2)/293.15,303.15/
DATA EIN; II/.44,3.1+1595//
DATA TO, FRANS/294.2,J.4711/
DATA TO, FRANS/294.2,J.4711/
DATA EO, AO/O.30,0.30/
DATA EO, AO/O.30,0.30/
DATA EO, AO/O.30,J.35/
DATA HEIGHT, SIDE, PATH/OJJJ.0.2197.0,112725.J/
DATA HEIGHT, SIDE, PATH/OJJJ.0.2197.0,112725.J/
DATA HEIGHT, SIDE, PATH/OJJJ.0.2203.0,112725.J/
DATA HEIGHT, SIDE, PATH/SOJO.0.2203.0,112725.J/
DATA HEIGHT, SIDE, PATH/SOJO.0.2203.0,112725.J/
DATA HEIGHT, SIDE, PATH/SOJO.0.2213.0,113196.0/
DATA HEIGHT, SIDE, PATH/SOJO.0.2213.0,113196.0/
DATA BEIGHT, SIDE, PATH/SOJO.0.1213.0.113196.0/
DATA PATHS, AREAS/193098.1,14+5030./
DATA PATHS, AREAS/193098.1,14+5030./
DATA PATHS, AREAS/193098.1,14+5030./
DATA PATHS, AREAS/193098.1,14+5030./
DATA PATHS, AREAS/193098.1,144200./
DATA PATHS, AREAS/193098.1,144200./
DATA PATHS, AREAS/193098.1,116000./
DATA PATHS, AREAS/19507.1,116000./
DATA APERT TROPT, RESPONTING TO THE PATHS AREAS/19507.1,116000./
DATA APERT TROPT, RESPONTING TO THE PATHS AREAS/19507.1,116000./
DATA CI,C2,SE/S.74152-1,21.4388,5.07E-025.30.,35.,40.,45.,50.,

DATA (ES(2,L),L=1,91.5)/0.-5.,10.,15.,20.,25.,30.,35.,40.,45.,50.,

DATA (ES(2,L),L=1,91.5)/0.-7.,74.5.79.3,413.0,444.1,409.6,493.4,

1 391.3,472.219.1,254.4.3300.3,342.2,379.3,413.0,444.1,409.6,493.4,

DATA (ES(2,L),L=1,91.5)/0.-5.,43.9.0,13.7,19.6,25.4,32.9,43.0,

DATA (ES(2,L),L=1,91.5)/0.-5.,43.9.0,13.7,19.6,25.4,32.9,43.0,

DATA (ES(2,L),L=1,91.5)/0.-5.,43.9.0,13.7,19.6,25.4,32.9,43.0,

DATA (ES(2,L),L=1,91.5)/0.-5.,43.9.0,13.7,19.6,25.4,32.9,43.0,

DATA (ES(2,L),L=1,91.5)/0.-5.,43.9.0,13.7,19.6,25.4,32.9,43.0,

DATA (ES(3,L),L=1,91.5)/0.-5.,43.9.0,13.7,19.6,25.4,32.9,43.0,

DATA (ES(3,L),L=1,91.5)/0.-5.,43.9.0,13.7,19.6,25.4,32.9,43.0,

DATA (ES(3,L),L=1,91.5)/0.-5.,43.9.0,13.7,19.6,25.4,32.9,43.0,

DATA (ES(3,L),L=1,91.5)/0.-5.,43.9.0,13.7,19.6,25.4,32.9,43.0,

DATA (ES(3,L),L=1,91.5)/0.-5.,43.9.0,13.7,19.6,25.4,32.9,43.0,

DATA (ES(2,L),L=1,91.5)/0.-5.43.9,94.2,36.3,102.1,105.7/

BELCO DATA ARE IN CM OR CM2
    CC
  C
  С
  00000 00000
                                    THE BELOW DATA ARE IN CM OR CM2
                                                                                     DATA #1,#1,#2,#3/1620.720.,210.,930./
DATA #4,#14,#5,#5,#7/900.,3cv.,150.,o18.,1260./
DATA #4,#10,%5,#7/900.,3cv.,150.,o18.,1260./
DATA #4,#10,%5,#7/900.,3cv.,150.,o18.,1260./
DATA #4,#10,%5,#7/900.,3cv.,150.,o18.,1260./
DATA #4,#10,%5,#7/900.,3cv.,150.,o18.,1260./
DATA #4,#10,%5,#7/900.,2197.,112389./
DATA PARIS,#REAS/164007.,1216c00./
DATA EM,T4/0.90,492.15/
DATA TRANS/J.425%

EATA (WE(N),N=1,40)/5.000,4.983,4.975,4.963,4.950,4.938,4.926,
14.910,4.902,4.390,4.78,4.660,4.254,4.7417,4.706,4.031,4.608,4.790,
14.765,4.773,4.762,4.751,4.735,4.728,4.717,4.706,4.031,4.608,4.673,
14.602,4.031,+640,4.630,4.619,4.634,4.598,4.587,4.577,4.560,4.536/
DATA (WE(N),N=41,52)/4.545,4.535,4.525,4.515,4.505,4.494,4.484,
14.474,4.4464,4.454,4.444,4.35/
EATA (T3L(N),H=1,33)/0.1577,0.1800,0.2070,0.2070,0.2070,0.22448,
10.2917,0.3342,0.3619,0.3797,0.3869,0.3962,0.507,0.3952,0.6623,0.5907,0.4242,
10.4747,0.5732,0.6230,0.647,0.0438,0.6244,0.343,0.6625,0.6931,
10.6880,0.0017,0.0505,0.0467,0.0305,0.5602,0.6643,0.6868,0.7053/,
DATA (T3L(N),#=1,33)/0.1577,0.1800,0.3905,0.5624,0.6443,0.6625,0.5907,0.5907,0.22448,
10.2917,0.3342,0.3619,0.3793,0.7265,0.77131,0.6099,0.627,0.5907,0.6217,0.6844,0.6344,0.634,0.634,0.634,0.634,0.636,0.3557,0.6217,0.6342,0.6644,0.634,0.634,0.634,0.634,0.634,0.634,0.634,0.634,0.634,0.634,0.634,0.634,0.634,0.634,0.634,0.634,0.634,0.634,0.634,0.634,0.634,0.634,0.634,0.634,0.634,0.634,0.634,0.634,0.634,0.634,0.634,0.634,0.634,0.634,0.634,0.634,0.634,0.634,0.634,0.634,0.634,0.634,0.634,0.634,0.634,0.634,0.634,0.634,0.634,0.634,0.634,0.634,0.634,0.634,0.634,0.634,0.634,0.634,0.634,0.634,0.634,0.634,0.634,0.634,0.634,0.634,0.634,0.634,0.634,0.634,0.634,0.634,0.634,0.634,0.634,0.634,0.634,0.634,0.634,0.634,0.634,0.634,0.634,0.634,0.634,0.634,0.634,0.634,0.634,0.634,0.634,0.634,0.634,0.634,0.634,0.634,0.634,0.634,0.634,0.634,0.634,0.634,0.634,0.634,0.634,0.634,0.634,0.634,0.634,0.634,0.634,0.634,0.634,0.634,0.634,0.634,0.634,0.634,0.634,0.634,0.634,0.634,0.634,0.634,0.634,0.634,0.634,0.634,0.634,0.634,0.634,0.634,0.634,0.6
      c
```

```
10.2046, J. 1438, J. 1127, J. 0124/
c
                        000
                  CALCULATION OF NORMAL COMPONENTS OF ES
                  DO 6 I=1,2

DC 5 K=1,3

DC 4 L=1,91

1H=D(L)*PI/180.

IF (I.20:1) A=JIN (IN)

IF (I.20:1) A=COS (IN)

ESN (I,K,L)=ES (K,L)*1

HDLE (I,K,L)=(9.54L-37,*ESN(I,K,L)

CONTINUE

CONTINUE
         6
               WRITE (6,7)
FORMAT (1', dx, 'SCLAR CONSTANT OF IN W/M2'///)
WRITE (6,5)
FORMAT (1X, 'ALL. ANGLE', 3X, 'UNGUSCUFED SUN', 2X, 'SUN WITH LIGHT CLCU
DS', 2X, 'SUN WITH HEAVY STOLM CLCUDS'/)
DO 10 L=1, 11

#RITE (6,7) D(L), (ES(K,L), F=1,3)
FORMAT (1X, FD.U, 5X, F10.2, 5X, F10.2, 14X, F10.2)
CCNTINU
WRITE (6,7) D(L), (ES(K,L), F=1,3)
FORMAT (1X, FD.U, 5X, F10.2, 5X, F10.2, 14X, F10.2)

WRITE (6,7) D(L), (ES(K,L), F=1,3)
FORMAT (1X, FD.U, 5X, F10.2, 5X, F10.2, 14X, F10.2)
WRITE (6,7) D(L), (ES(K,L), F=1,3)
FORMAT (1X, FD.U, 5X, F10.2, 5X, F10.2, 14X, F10.2)
WRITE (6,7)
WRITE (6,7)
WRITE (6,7)
WRITE (6,7)
WRITE (6,7)
WRITE (6,7)

WRITE (6,7)

WRITE (6,7)

WRITE (6,7)
\alpha
         7
          8
       10
       11
       12
                   13
       14
       15
16
                   DO 20 I=1,2
DC 19 J=1,2
DO 18 K=1,3
DO 17 L=1,)1
TO4=TO**4.
TIN4=TIN(J)**4.
ESN(I,K,L)=(L./J.)*ESN(I,K,L)
A=ESN(I,K,L)*AO**3O**TO4**EIN**SB**TIN4
d=(20+c13)*50
XK=(A/E)**0.20
T4=AK**4.
F1=EO*SB**(T4-TO4)*EIN*SB**(T4-TIN+)*HCGNV**(XK-IO)-ESN(I,K,L)*AC
XA=XK+F1
X4=XA**4.
    2000
                             22-20*53* (X4-T04) +ELN*56* (X4-TIA+) +HCCHV* (XA-T0) -ESN (I,X,I) *AC

XK1=XK-F1*F1/(F2-F1)

DIF=ASS (XK1-KK)

XK=XK1
```

```
IF (DIF.GT.000001) GC IG 2000 T(I,J,K,L) = YK CONTINUE CONTINUE CONTINUE CONTINUE CONTINUE
        17
18
19
20
DO 27 J=1,2
W31IE(6,21) TIN(J)
FORMAT(11,3X,'SHIP BODY TEMPERATURE IN K FOR TIN=',F7.2,11,

WEITE(5,22)
FORMAT(0,9X,'UNOBSCURED SUN',JX,'SUN WITH LIGHT CLCOD',2X,

WBITE(5,23)
FORMAT(1X,'ALTITUDZ',1X,3('HGRIZUNTAL',2X,'VERFICAL',2X))
WBITE(5,24)
FORMAT(1X,'AMGLE',4X,3('FACE',6X,'FACE',6X)//)
DO 26 L=1,11
WAITE(5,25) D(L), (T(L,J,1,L),L=1,2), (T(L,J,2,L),L=1,2),

FORMAT(1X,F6.1,3(F10.2,1X,F10.2,1X))
CONTINUE
CONTINUE
        21
        22
        23
        24
                        CALCULATION OF NORMAL COMPONENT OF SPECTRAL BADIANCE IN THE WINDOW OF 5 MICROMETER (I.E. 4.435-5.000)
                       DC J2 I=1,2
DO 51 J=1,2
DO 30 K=1,3
DO 29 L=1,91
S1=0.0
S2=0.0
X1=C2/(T(I,J,K,I)*4.435E-04)
Y2=C2/(T(I,J,K,I)*5.000E-04)
DC J2 =1,3
A1=FLOAT(1)*X1
A2=FLOAT(1)*X2
B=FLOAT(1)*X2
B=FLOAT(1)*X2
B=FLOAT(1)*X2
B=S1+EAP(-AI)*(A1**3.+3.*A1**A1+6.*A1+6.)/B
S2=S2+EXP(-AI)*(A2**3.+3.*A2**A2*6.*A2+6.)/B
CCNTINUE
HDLT(I,J,K,I)=C1*EC*(T(I,J,K,I)**4.)*(S2-S1)/(PI*(C2**4.))
HDL(I,J,K,I)=HDLT(I,J,K,I)+HDLE(I,K,I)
CONTINUE
CONTINUE
CONTINUE
         23
         29
30
31
32
 0000
                         CALCULATION OF NORMAL COMPONENT OF SEA RADIANCE IN THE WINDOW OF 5 MICECMETER (I.E. 4.435-5.000)
                                   S1=0.0

S2=0.0

X1=C2/(TW*4.435E-04)

X2=C2/(TW*5.000E-04)

D0 128 M=1.8

A1=FLOAT(M) *X1

A2=FLOAT(M) *X2

B=PLOAI(A) **4.

S1=51+6XP(-A1) *(A1**3.+3.*A1*A1+6.*A1+6.)/B

S2=S2+EX9(-A2) *(A2**3.+3.*A2*A2+6.*A2+6.)/B

CONTINUE

HDIM=C1*EW*(TW**4.) *(S2-S1)/(FI*(C2**4.))
      128
  000000
                      DO 40 J=1,2
WRITE(6,33)
PORMAI(11,1X,*NORMAL COMPONENT OF RADIANCE IN W/ (CM2*SB) *//1X,
1*DUE TO THERMAL MADIATION*/)
          33
```

```
WRITE (0,34) fin (J) FORMAT(2X, THE WINDOW OF 5 SICHCHETER FOR TIM=',F7.2,1X,'K'//) WRITE (6,35) FORMAT(1,000) FOR
3 3 5 6 3 7 8990 8 CFR
                                            WRITE (0,30)
WRITE (0,30)
FORMAT (1X, 'ALTITUDE',1X,3 ('UGRIZONTAL',2X,'VERTICAL',2X))
WRITE (6,37)
FORMAT (1X, 'ANGLE',4X,3 ('FACE',8X,'FACE',6X)//)
DO 39 L=1,91
WRITE (0,33) D(L), (HDL(I,J,1,L), I=1,2), (HDL(I,J,2,L), I=1,2),
(HDL(I,J,3,L), I=1,2)
FORMAT (1X,Fo.1,24,3 (E10.4,1X,E10.4,1X))
CONTINUE
CONTINUE
                                                      WRITE(6,138) TW, HDLW
PORMAT(*1*,2x, 'SEA RADIANCE FOR TW=*, F6.2,2x,E15.4)
                         CALCULATION OF SHIP SURFACES AND THEIR COURESPONDING MEAN ELEVATIONS PROM THE SEA, INSIDE THE INSTANTANEOUS DETECTOR FIELD OF VIEW IN THE REGION OF THE HIGHER RADIANT INTENSITY.
                                                A1=N1*H1
A11=(SIDE-J1)*H1
HM1=H1/2.
  C
                                                A2=41*H2
A22=(SIDE-W1) *H2
HM2=H1
  C
                                                A3=SIDE*H3
HM3=H1+H3/2.
   C
                                                 HM4=H1+H3+H4/2.
    С
                                                  A5=95*85
                                                 BM5=H1+H3+H4+H5/2.
   С
                                                HM6=H1+HJ+H4+d5
    С
                                                 A7=SIDE*H7-PI*#4*#4/4.
EM7=H1+E3
   00000
                                 CALCULATION OF THE BADIANT INTENSITY IN (#/SR) OF THE SURFACES (A1+A11), (A2+A22), A3, A4, A5, A7 IN THE WINDOW OF 5 MICROMETER, IN THE DIFFECTION SHIP-MISSILE
                                            MICROMETER, IN THE DIFFECTION SHIP—SISSING AND 42 K=1,3

DO 41 L=1,91

RIN=HDL(2,2,K,L) *A1+dDL(2,1,K,L) *A11

C1=(1.-C11*C11) **0.5

RI1(K,L) = BIIN*C1

RI2N=HDL(1,2,K,L) *A2+HDL(1,1,K,L) *A22

C22=(HEIGHI-HM2)/PATH

RI2N=HDL(2,1,K,L) *A3

C33=(HEIGHI-HM3)/PATH

C3=(1.-C3*C35)**0.5

BI3(K,L) = BI3N*C3

BI4N=HDL(2,2,K,L) *A4

C44=(HEIGHI-HM4)/PAIH

C44=(1.-C44*C44)**0.5

RI4(K,L) = RI4N*C4

RI5N=HDL(2,2,K,L) *A5

C55=(HEIGHI-HM4)/PAIH

C4=(1.-C55*C55)**0.5

RI4(K,L) = RI4N*C4

RI5N=HDL(2,2,K,L) *A5

C55=(HEIGHI-HM5)/PAIH

C5=(1.-C55*C55)**0.5

RI6(K,L) = RI5N*C5

C66=(HEIGHI-HM6)/PAIH

RI6=BI6H*C66

BI7N=HDL(1,1,K,L) *A7
```

```
CC CALCULATION OF NGISE VOLTAGE

CALCULATION OF NGISE VOLTAGE

CALCULATION OF NGISE VOLTAGE

CALCULATION OF NGISE VOLTAGE

CONTINUE

CALCULATION OF NGISE VOLTAGE

CALCULATION OF NGISE VOLTAGE

OMINST=SIDE*SIDE*(FATH*FATH)

AD=DSINST*FOCIT*FOCIT*FOCIT*FOCIT*FOCIT*FOCIT*FOCIT*FOCIT*FOCIT*FOCIT*FOCIT*FOCIT*FOCIT*FOCIT*FOCIT*FOCIT*FOCIT*FOCIT*FOCIT*FOCIT*FOCIT*FOCIT*FOCIT*FOCIT*FOCIT*FOCIT*FOCIT*FOCIT*FOCIT*FOCIT*FOCIT*FOCIT*FOCIT*FOCIT*FOCIT*FOCIT*FOCIT*FOCIT*FOCIT*FOCIT*FOCIT*FOCIT*FOCIT*FOCIT*FOCIT*FOCIT*FOCIT*FOCIT*FOCIT*FOCIT*FOCIT*FOCIT*FOCIT*FOCIT*FOCIT*FOCIT*FOCIT*FOCIT*FOCIT*FOCIT*FOCIT*FOCIT*FOCIT*FOCIT*FOCIT*FOCIT*FOCIT*FOCIT*FOCIT*FOCIT*FOCIT*FOCIT*FOCIT*FOCIT*FOCIT*FOCIT*FOCIT*FOCIT*FOCIT*FOCIT*FOCIT*FOCIT*FOCIT*FOCIT*FOCIT*FOCIT*FOCIT*FOCIT*FOCIT*FOCIT*FOCIT*FOCIT*FOCIT*FOCIT*FOCIT*FOCIT*FOCIT*FOCIT*FOCIT*FOCIT*FOCIT*FOCIT*FOCIT*FOCIT*FOCIT*FOCIT*FOCIT*FOCIT*FOCIT*FOCIT*FOCIT*FOCIT*FOCIT*FOCIT*FOCIT*FOCIT*FOCIT*FOCIT*FOCIT*FOCIT*FOCIT*FOCIT*FOCIT*FOCIT*FOCIT*FOCIT*FOCIT*FOCIT*FOCIT*FOCIT*FOCIT*FOCIT*FOCIT*FOCIT*FOCIT*FOCIT*FOCIT*FOCIT*FOCIT*FOCIT*FOCIT*FOCIT*FOCIT*FOCIT*FOCIT*FOCIT*FOCIT*FOCIT*FOCIT*FOCIT*FOCIT*FOCIT*FOCIT*FOCIT*FOCIT*FOCIT*FOCIT*FOCIT*FOCIT*FOCIT*FOCIT*FOCIT*FOCIT*FOCIT*FOCIT*FOCIT*FOCIT*FOCIT*FOCIT*FOCIT*FOCIT*FOCIT*FOCIT*FOCIT*FOCIT*FOCIT*FOCIT*FOCIT*FOCIT*FOCIT*FOCIT*FOCIT*FOCIT*FOCIT*FOCIT*FOCIT*FOCIT*FOCIT*FOCIT*FOCIT*FOCIT*FOCIT*FOCIT*FOCIT*FOCIT*FOCIT*FOCIT*FOCIT*FOCIT*FOCIT*FOCIT*FOCIT*FOCIT*FOCIT*FOCIT*FOCIT*FOCIT*FOCIT*FOCIT*FOCIT*FOCIT*FOCIT*FOCIT*FOCIT*FOCIT*FOCIT*FOCIT*FOCIT*FOCIT*FOCIT*FOCIT*FOCIT*FOCIT*FOCIT*FOCIT*FOCIT*FOCIT*FOCIT*FOCIT*FOCIT*FOCIT*FOCIT*FOCIT*FOCIT*FOCIT*FOCIT*FOCIT*FOCIT*FOCIT*FOCIT*FOCIT*FOCIT*FOCIT*FOCIT*FOCIT*FOCIT*FOCIT*FOCIT*FOCIT*FOCIT*FOCIT*FOCIT*FOCIT*FOCIT*FOCIT*FOCIT*FOCIT*FOCIT*FOCIT*FOCIT*FOCIT*FOCIT*FOCIT*FOCIT*FOCIT*FOCIT*FOCIT*FOCIT*FOCIT*FOCIT*FOCIT*FOCIT*FOCIT*FOCIT*FOCIT*FOCIT*FOCIT*FOCIT*FOCIT*FOCIT*FOCIT*FOCIT*FOCIT*FOCIT*FOCIT*FOCIT*FOCIT*FOCIT*FOCIT*FOCIT*FOCIT*FOCIT*FOCIT*FOCIT*FOCIT*FOCIT*FOCIT*FOCIT*FOCIT*FOCIT*FOCIT*FOCIT*F
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